

Hypersonic Shock/Boundary-Layer Interaction Database: New and Corrected Data

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Introduction

As described fully in Ref. 1, an effort was begun in 1989 at the Penn State University Gas Dynamics Laboratory to perform a critical review of the available hypersonic data and to assemble a selected database for purposes of CFD code validation and turbulence modeling. The effort was sponsored by the NASP Program through NASA-Ames Research Center, and was a part of an overall task to develop compressible turbulence models. Ref. 1, a database report on hypersonic shock wave/turbulent boundary-layer interactions, was the product of phase 1 of that effort. Phase 2 produced a similar database, reported in Ref. 2, covering the topics of attached hypersonic boundary layers in pressure gradients and compressible turbulent mixing layers. The present report represents the result of the third and final phase, namely, recent additions and corrections to the hypersonic shock wave/turbulent boundary-layer interaction database originally given in Ref. 1.

The new datasets included here are those which have come to our attention since the completion of Phase 1 of the database effort at the end of 1990, and which were able to pass the acceptance criteria originally applied in Phase 1. Those criteria are listed by name below, but the reader is directed to Ref. 1 for details of their application:

(NECESSARY CRITERIA)

- 1) BASELINE APPLICABILITY
- 2) SIMPLICITY
- 3) SPECIFIC APPLICABILITY
- 4) WELL-DEFINED EXPERIMENTAL BOUNDARY CONDITIONS
- 5) WELL-DEFINED EXPERIMENTAL ERROR BOUNDS
- 6) CONSISTENCY CRITERION
- 7) ADEQUATE DOCUMENTATION OF DATA
- 8) ADEQUATE SPATIAL RESOLUTION OF DATA

(DESIRABLE CRITERIA)

- 1) TURBULENCE DATA
- 2) REALISTIC TEST CONDITIONS
- 3) NON-INTRUSIVE INSTRUMENTATION
- 4) REDUNDANT MEASUREMENTS
- 5) FLOW STRUCTURE AND PHYSICS

Discussion of New Data

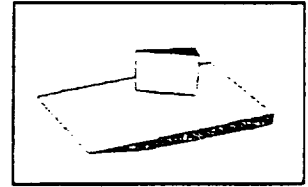
Ref.: 3-6

Author: Kussoy and Horstman

Geometry: 3-D Fin

Mach number: 8.2

Data: p_{wall} , q_{wall} , c_f , flowfield pitot surveys



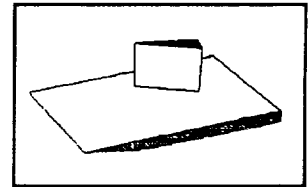
Ref.: 7-9

Author: Rodi and Dolling

Geometry: 3-D Fin

Mach number: 4.9

Data: p_{wall} , q_{wall} , surface-flow traces



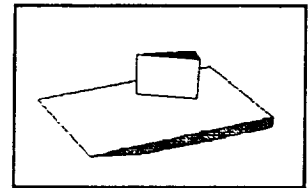
Ref.: 10-11

Author: Lee and Settles

Geometry: 3-D Fin

Mach number: 3, 4

Data: c_h



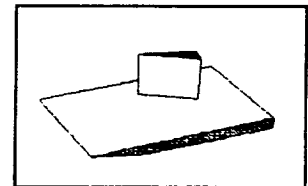
Ref.: 12-13

Author: Hsu and Settles

Geometry: 3-D Fin

Mach number: 3, 4

Data: flowfield density maps



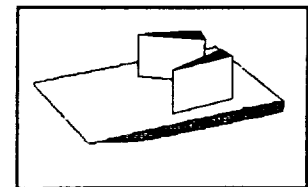
Ref.: 14

Author: Kussoy and Horstman

Geometry: Crossing Oblique Shock Waves

Mach number: 8.3

Data: p_{wall} , q_{wall} , flowfield pitot surveys



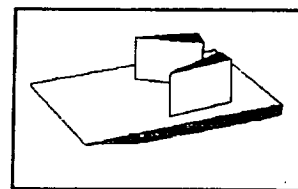
Ref.: 15-17

Author: Garrison and Settles

Geometry: Crossing Oblique Shock Waves

Mach number: 4

Data: p_{wall} , c_f , flowfield pitot surveys



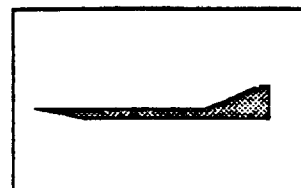
Ref.: 18-20

Author: Kuntz *et al.*

Geometry: 2-D Compression Corner

Mach number: 3

Data: p_{wall} , mean & fluctuating flowfield surveys (2-channel LDV)



The new data since 1991 can be characterized in 3 groups as follows: 1) Mach 8 single- and double-fin (crossing-shock) interactions, 2) more advanced measurements of supersonic single- and double-fin interactions, and 3) more advanced measurements of a supersonic 2-D compression corner.

The new Mach 8 data are the result of a concerted experimental program by Kusoy and Horstman in the NASA-Ames Research Center 3.5-foot hypersonic wind tunnel (Refs. 3-6 and 14). These carefully-documented datasets fulfill several of the needs pointed out in Ref. 1 and its companion technical paper (Ref. 21), including Mach number firmly within the hypersonic regime, more-complex interaction types, and more emphasis on 3-D interactions.

The new supersonic single-fin-interaction datasets are the results of research programs at Penn State (Refs. 10-13) and the University of Texas-Austin (Refs. 7-9) aimed at providing advanced data for code validation and turbulence modeling. They feature heat transfer data to supplement the skin friction data already including in Ref. 1, and flowfield density maps obtained non-intrusively by conical holographic interferometry.

The new supersonic double-fin or crossing-shock interaction data (Refs. 15, 16) feature surface pressures, skin friction coefficients, and one plane of flowfield pitot-pressure data for this important inlet-type interaction. Readers interested in this interaction should also see Refs. 14 and 22-23.

Finally, the new 2-D compression corner data listed here (Refs. 18-20) were actually available previously, and were included in the literature search of Ref. 1, but, due to an oversight, were not subjected to an evaluation. That was unfortunate, since this dataset provides valuable LDV data to supplement the mean-flow and hot-wire data previously available for supersonic compression corners.

However, the availability of Mach 3 LDV compression corner data in Refs. 18-20 creates the following dilemma: The LDV magnitudes of Reynolds stresses in these interactions are 2 to 4 times larger than the levels found using hot-wire anemometry by Smits *et al.*, Refs. 1 and 24) in a similar experiment. The LDV authors speculate (see discussion just prior to Conclusions of Ref. 18) that this discrepancy is due to inherent errors

in slanted-hot-wire calibrations. However, it is not reasonable to discount the hot-wire data on the basis of this speculation alone. Moreover, discussions with D. W. Kuntz concerning the LDV dataset appear to confirm that the higher LDV values of Reynolds stress are not simply due to shock unsteadiness, since pdf plots of the data do not reveal bimodal distributions. Though the hot-wire and LDV experiments were conducted in different test facilities varying in size by a factor of two, the test conditions were similar enough that the discrepancy cannot be thus explained. Given such an unresolved discrepancy, we have decided to retain both sets of data in the present database with the following warning:

Note: One or both of the Smits (Ref. 24) and Kuntz (Ref. 18) 2-D compression corner datasets is incorrect insofar as the levels of turbulent Reynolds stresses are concerned. It is not presently possible to determine where the error lies. Thus both experiments are included in the database in order to show possible limits on Reynolds stresses in such interactions, or possible error levels in the experiments. Beyond that, there is only the time-honored disclaimer, *caveat emptor*.

Discussion of Corrected Data

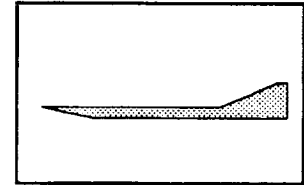
Ref.: 25, private communication

Author: Zheltovodov, A. A., *et al*

Geometry: 2-D Compression Corner

Mach number: 3

Data: p_{wall} , c_h , mean and fluctuating flowfield surveys (pitot and hot-wire anemometry)



Two small but significant errors were found in this dataset as it is presented in Ref. 1. First, although the step height $h = 15$ mm, is correctly given for the mean-flow test geometry shown on page 108 of Ref. 1, no value is given for h in the case of the heat transfer model shown on the next page. In fact, $h = 6$ mm is the correct value for the heat transfer model.

Second, in the tabulated data on page 119 of Ref. 1 and in the data file ZHELT.DAT, a typographical error appears. In the heading:

```
*****HeatTransferDataonModelFFS25*****
M1 = 3.01,  $\alpha_1 = 280$  W/m**2 K
```

The correct value of the incoming flat-plate heat transfer rate α_1 should be 180 W/m²K. This correction has been verified by e-mail correspondence with A. A. Zheltovodov. The diskette accompanying the present report contains a corrected data file designated ZHELT2.DAT.

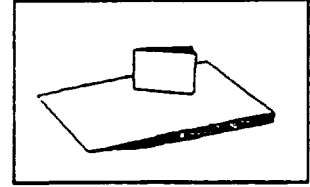
Ref.: 26

Author: Kim, K-S, *et al*

Geometry: 3-D Fin

Mach number: 3, 4

Data: p_{wall} , c_f , surface-flow angles



Based on a suggestion by D. J. Monson, the laser-skin-friction data reduction scheme of Kim (Ref. 26) was re-examined by Garrison (Ref. 17). It was found that, in the case of high peak skin friction levels in swept interactions where few laser interference fringes were available, Kim's approach led to a significant overestimate of the skin friction coefficient (Ref. 27). Garrison then repeated experiments at the peak points of Kim's strongest single-fin interaction (Mach 4, $\alpha = 20$ degrees) and found lower skin friction values than those originally found by Kim. Kim's original skin friction distribution for this case, as tabulated in Ref. 1, is:

MACH 4, ALPHA = 20 DEGREES

BETA	CF	ERROR BAR
------	----	-----------

56.00	9.868E-4	2.774E-5
-------	----------	----------

51.00	9.835E-4	4.131E-5
-------	----------	----------

48.80	1.114E-3	4.380E-5
-------	----------	----------

44.00	1.199E-3	4.645E-5
-------	----------	----------

40.50	1.820E-3	8.979E-5
-------	----------	----------

39.50	1.655E-3	3.913E-5
-------	----------	----------

31.00	2.649E-3	9.820E-5
-------	----------	----------

26.30	9.494E-3	6.718E-4
-------	----------	----------

23.30	7.733E-3	5.920E-4
-------	----------	----------

21.00	5.749E-3	4.377E-4
-------	----------	----------

With Garrison's correction of the 8th and 9th data points in this table, the corrected skin friction distribution is:

MACH 4, ALPHA = 20 DEGREES

BETA	CF	ERROR BAR
------	----	-----------

56.00	9.868E-4	2.774E-5
-------	----------	----------

51.00	9.835E-4	4.131E-5
-------	----------	----------

48.80	1.114E-3	4.380E-5
-------	----------	----------

44.00	1.199E-3	4.645E-5
-------	----------	----------

40.50	1.820E-3	8.979E-5
-------	----------	----------

39.50	1.655E-3	3.913E-5
-------	----------	----------

31.00	2.649E-3	9.820E-5
-------	----------	----------

26.50	5.070E-3	5.000E-4
-------	----------	----------

22.00	5.510E-3	5.000E-4
-------	----------	----------

21.00 5.749E-3 4.377E-4

A corrected data file with the name KIM2.DAT has been included on the diskette accompanying the present report.

This error is an unfortunate one, since it changes the conclusions of Ref. 26. In fact, the computational solutions described there are in much better agreement with the corrected data than they were with the original, erroneous data. However, an examination of this dataset indicates that such peak-skin-friction errors occurred only in the strongest interaction (Mach 4, $\alpha = 20$ degrees), so that the results given in Refs. 1 and 26 and the conclusions drawn in ref. 26 for the weaker interactions are still believed to be valid.

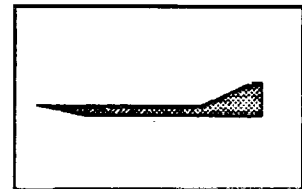
Ref.: 28, 29

Author: Smits, A. J., *et al*

Geometry: 2-D Compression Corner

Mach number: 3

Data: p_{wall} , c_f , mean & fluctuating flowfield surveys (pitot and hot-wire anemometry)



Since the publication of Ref. 1, several computational groups have made use of the mean-flow component of this dataset, which was tabulated and appeared in the ASCII file named SETTLES.DAT which was on the diskette included with Ref. 1. The dataset has thus had a rather thorough "workout," and several errors and problems were found as follows.

To put the mean-flow dataset in perspective, note that it consists of 1970's-vintage 2-D compression corner data taken by Settles *et al*. This work was originally reported in Refs. 30-33, which contain some discrepancies attributable to the original experimenters as described below. It was also submitted as a test case for the 1980-81 AFOSR-HTTM-Stanford Conference on Complex Turbulent Flows, whence Ref. 28 was prepared. Ref. 28 is still a complete and almost-up-to-date discussion and tabulation of the dataset, but this reference is, by its nature, not widely available. Unfortunately, the later tabulation of the same dataset for Ref. 29 involved re-reduction of the raw data by persons other than the original experimenters, whereupon additional discrepancies occurred. An attempt is made here to resolve these issues.

1) Tabulation of Freestream and Incoming-Flow Conditions

Ref. 1 contains no proper tabulation of freestream and incoming-flow conditions for the four compression-corner cases included in the dataset. This information, from Ref. 28, is now included in the present data file SETTLES2.DAT and in the printed tabulation reproduced in this report, along with a list of definitions of terms. It should be noted that minor revisions of some of the incoming conditions, especially PINF values, have been made on account of a re-examination of the dataset after Ref. 28 was prepared.

2) Correction of Wall-Pressure Distributions

The manner in which the wall pressures were presented in Ref. 1 was inconsistent, confusing, and erroneous in places. The revised distributions are presented in terms of P_{wall}/P_{INF} , with P_{INF} tabulated nearby, and have been gone over carefully for errors.

3) Correction of Confusion Surrounding Skin-Friction Distributions

By far the worst problem with this dataset as presented in Ref. 1 concerns the skin friction distributions. The problem arose because the original experimenters chose to present the skin friction coefficient in two different forms: wall shear stress normalized by $\rho_{\infty} u_{\infty}^2$ ("CFINF" in Ref. 1) and normalized by $\rho_{edge} u_{edge}^2$ ("CF" in Ref. 1). In retrospect, the former form is the proper skin friction coefficient, while the latter is confusing and essentially worthless. This unfortunate situation has been cleared up in the present tabulation and SETTLES2.DAT file by deleting the "CF" column and renaming the "CFINF" column "CF," which is now unequivocally defined as the wall shear stress normalized by $\rho_{\infty} u_{\infty}^2$.

To make matters worse, early publications such as Ref. 33, due to a misunderstanding between experimental and computational authors, compared properly-defined computed skin friction distributions with the above improperly-defined experimental values. Thus Figs. 3 and 9 of Ref. 33 show data points which are significantly smaller than they should be. This mistake was later discovered and corrected, but the confusion factor still exists in the early literature. Hopefully the present report will clear up this confusion.

4) Mean Profile Corrections

A recheck of the mean flowfield profiles of this dataset uncovered some minor errors and needed clarifications. These have been corrected and implemented in the present SETTLES2.DAT file.

Closure

The publication of this report brings to an end the effort by the Penn State Gas Dynamics Lab, under NASA-Ames support, to establish a database for hypersonic boundary layers, interacting flows, and compressible mixing. There remains only the necessity to comment on the limitations of the present database for users of advanced turbulence models.

In general, none of the experiments included in this database contains enough information to properly specify the boundary conditions of any *turbulent-flow* computation. Assumptions must always be made. As an example of a reasonable assumption, Horstman (Ref. 34) carries out a finite-difference boundary-layer calculation for the test surface ahead of a shock/boundary-layer interaction. At the point where the computed boundary-layer displacement thickness matches that of the experiment, Horstman uses the computed boundary-layer turbulence quantities as input conditions. A similar approach is adopted by Morrison *et al.* (Ref. 35), who attempt to find a location where all three computed boundary-layer thicknesses and the skin friction coefficient are within 15% of the experimentally-determined values.

Both these approaches are reasonable, and similar schemes may work as well, if it is established that the incoming turbulent boundary-layer is in equilibrium in the experiment. However, in shock interaction cases where the incoming boundary-layer is not in equilibrium (of which there are none in the present database), much more documentation and detail of the incoming boundary layer would be required in the experiment in order to provide proper information for a computational simulation.

References

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Appendix: Data Tabulation

There follows a tabulation of pertinent data from the new and corrected datasets which make up the hypersonic shock/boundary-layer interaction database. For each dataset, a brief discussion of the data is given for the benefit of users. However, users are strongly encouraged to consult the original references for more detail on what was measured and how it was accomplished. Similarly, no attempt has been made to tabulate all available data from each of these studies, but rather only those data most pertinent to the issues of turbulence modeling and code validation. In several cases, additional data may be had from the original publications. Moreover, of the data selected for inclusion, only initial profiles and samples are printed in this report, since brevity is required and paper tabulations are no longer of much use when machine-readable data are readily available. A 3.5" double-sided high-density diskette is also provided in original copies of this report. This disk contains the complete data-tables of this Appendix in machine-readable ASCII files, formatted for MS-DOS computers. Individual ASCII files are given for each of the datasets, with filenames keyed to first authors as follows:

New Datasets:

GARRISON.DAT
HSU.DAT
KUNTZ.DAT
KUSSOY3.DAT (single-fin)
KUSSOY4.DAT (crossing-shocks)
LEE.DAT
RODI.DAT

Corrected Datasets:

SETTLES2.DAT

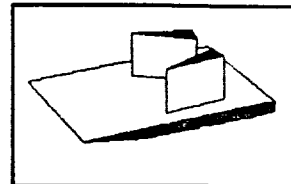
Ref.: 15-17

Author: Garrison and Settles

Geometry: Crossing Oblique Shock Waves

Mach number: 3 and 4

Data: p_{wall} , c_f , flowfield pitot surveys



The experiments were carried out using a double-fin test geometry mounted on a flat plate in the supersonic wind tunnel facility of the Penn State Gas Dynamics Laboratory. The flat-plate boundary-layers at Mach 3 and 4 are the same as those previously documented in Ref. 1(KIM.DAT). These are equilibrium turbulent boundary-layers developing naturally on a flat-plate at high Reynolds number, and are essentially adiabatic.

Opposing fins produced oblique shocks of opposite families, which intersected along the centerline of the flat plate. All test geometries were symmetric about this centerline. Fin angles-of-attack of 7x7, 9x9, 11x11, and 13x13 degrees were tested at Mach 3 and 4. A 15x15 degree case was also tested at Mach 4.

All data are described by a right-handed Cartesian x,y,z coordinate system. The origin of coordinates is on the centerline of the flat plate at the location of the fin leading-edges, ie 21.3 cm downstream of the flat-plate leading-edge. The x and z coordinates lie in the plane of the flat plate when $y = 0$, and in planes parallel to but above it for $y > 0$. The z coordinate is positive in the downstream direction, while the x coordinate is positive to the right of the plate centerline when viewed from the downstream direction. The z and x-locations of points within a given interaction are normalized by a reference "incoming" boundary-layer thickness, which is taken as $\delta_0 = 3.5$ mm for all test conditions.

The data file GARRISON.DAT contains tabulations of measured wall static pressures, skin friction distributions, and a single pitot-survey plane in these interactions. The wall static pressures were measured only on the centerlines of the symmetric interactions studied. Skin friction data, obtained by way of a laser-based technique (Ref. 17) were measured both on the centerline and on certain spanwise "cuts" at specific streamwise locations denoted by z/δ_0 values cited in the data file. The flowfield pitot survey data were obtained in a single x-y plane located at $z/\delta_0 = 32.33$ in the Mach 4, 15x15 deg interaction. While the flowfield features in this plane are discussed in Refs. 15, 22, and 23, the procedure of these flowfield surveys has not yet been described elsewhere in print. Interested readers should consult Ref. 16 when it becomes available.


```

*****
GARRISON.DAT
*****
SURFACE PRESSURE, SKIN FRICTION AND PITOT SURVEY DATA
SYMMETRIC CROSSING-SHOCK/TURBULENT BOUNDARY-LAYER
INTERACTIONS AT MACH 3 AND 4
*****
NOMENCLATURE & DEFINITIONS
*****
Cf = SKIN FRICTION COEFFICIENT
Z0 = INITIAL REFERENCE BOUNDARY LAYER THICKNESS AT L.E. LOCATION
    (DEFINED AS 3.5mm FOR ALL CASES)
H = FIN HEIGHT (= 8.25 cm)
P0 = FREESTREAM STAGNATION PRESSURE
Pw = WALL STATIC PRESSURE
Pinf = FREESTREAM STATIC PRESSURE
pt2 = PITOT PRESSURE
Re = REYNOLDS NUMBER
T0 = FREESTREAM STAGNATION TEMPERATURE
Tw = WALL TEMPERATURE
Taw = ADIABATIC WALL TEMPERATURE
W = WIDTH BETWEEN FIN LEADING EDGES (= 9.4 cm)
X = SPANWISE COORDINATE ORIGINATING ON TUNNEL CENTERLINE
Y = VERTICAL COORDINATE ORIGINATING ON FLAT PLATE
Z = STREAMWISE COORDINATE ORIGINATING AT FIN LEADING
    EDGE POSITION (21.3 cm DOWNSTREAM OF PLATE L.E.)
*****
INCOMING AND FREESTREAM FLOW CONDITIONS
*****
MACH 3
P0 = 750 kPa
Pinf = 20.4 kPa
T0 = 295 K
Re/m = 6.2E+07
Tw/Taw = 1.06 (roughly adiabatic)
*****
MACH 4
P0 = 1500 kPa
Pinf = 99 kPa
T0 = 295 K
Re/m = 6.8E+07
Tw/Taw = 1.06 (roughly adiabatic)
*****
INCOMING BOUNDARY LAYER PARAMETERS AT X=178 mm FROM FLAT-PLATE
LEADING EDGE
*****
Mach 3 (actual freestream Mach number was 2.91 for this survey)
Delta = 3.02 mm
Delta* = 0.895 mm
Theta = 0.184 mm
Ret(theta) = 9751
*****
Mach 4 (actual freestream Mach number was 3.88 for this survey)
Delta = 2.87 mm
Delta* = 0.950 mm
Theta = 0.128 mm
Ret(theta) = 9082
*****
CENTERLINE WALL PRESSURE DISTRIBUTIONS AT MACH 3
*****
z/80 13x13deg 11x11deg 9x9 deg 7x7 deg
      p/pinf p/pinf p/pinf p/pinf
*****

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8.9989 0.98581 1.0117 0.98868 1.0094
13.353 1.0402 1.0466 1.0279 1.0421
14.079 1.1602 1.0538 1.0259 1.0492
15.349 1.8911 1.1163 1.0329 1.0494
16.619 2.3104 1.4548 1.1263 1.1021
17.889 2.5082 1.8144 1.2572 1.1076
19.703 2.7039 2.1856 1.5945 1.238
21.429 2.7521 2.2983 1.7418 1.3429
22.969 3.0462 2.4007 1.8828 1.4387
24.239 3.1559 2.6327 2.1465 1.5298
26.779 3.5511 2.8183 2.3665 1.6709
28.049 4.1086 2.9662 2.4964 1.8918
29.319 4.6118 3.0773 2.5828 1.9614
30.589 4.7936 3.1915 2.6621 2.0121
32.403 5.1768 3.3939 2.7606 2.1142
33.129 5.2627 3.5051 2.7924 2.1403
34.399 5.5767 3.6958 2.8237 2.176
35.669 5.8293 3.8576 2.8857 2.2187
36.395 5.8524 3.905 2.9237 2.2262
39.479 5.2015 4.5838 3.0998 2.3517
41.475 4.0826 5.0753 3.1133 2.3967
43.289 2.8807 4.7789 3.0665 2.4035
44.559 2.5063 4.2625 3.0881 2.4294
45.829 2.2481 4.0106 3.5857 2.5234
47.099 1.856 3.2963 3.367 2.5321
48.369 1.7024 3.0198 3.4325 2.5874
49.639 1.6198 2.554 3.0653 2.6657
50.909 1.2915 2.2812 2.8738 2.7229
52.179 1.1545 2.0177 2.5945 2.8016
53.449 0.86939 1.4938 1.9485 2.7956
54.719 0.91548 1.5216 2.0365 2.8324
55.989 1.0919 1.5773 1.9042 2.8415
57.259 0.67488 1.0862 1.463 2.5718
58.529 0.64068 1.0748 1.3134 2.4989
*****
CENTERLINE WALL PRESSURE DISTRIBUTIONS AT MACH 4
*****
z/80 15x15deg 13x13deg 11x11deg 9x9 deg 7x7 deg
      p/pinf p/pinf p/pinf p/pinf p/pinf
*****
8.9989 1.0222 1.0521 1.0414 1.017 1.0348
10.269 0.9665 1.0342 1.0223 0.9968 1.0149
11.539 0.9395 1.032 1.0234 0.9901 1.0244
13.353 1.0421 1.0425 1.0286 1.0061 1.0147
14.079 1.294 1.0385 1.0213 0.9964 1.0076
15.349 2.1077 1.1262 1.0234 0.9862 1.0341
16.619 2.4778 1.6252 1.1204 1.0193 0.98821
17.889 2.6778 2.0441 1.3119 0.9908 1.0496
19.703 2.8588 2.4136 1.7294 1.1418 1.0794
20.429 2.9773 2.5618 1.9245 1.2605 1.0901
21.699 3.1775 2.6111 2.0294 1.3798 1.1642
22.969 3.7023 2.8273 2.2436 1.5371 1.2685
24.239 3.6624 2.8554 2.3429 1.6749 1.4579
26.779 4.9227 3.2875 2.5475 1.8806 1.5441
28.049 5.7257 3.7931 2.7016 2.0213 1.66
29.319 6.6439 4.2784 2.8324 2.1385 1.7643
30.589 7.1743 4.7166 3.0702 2.2943 1.9303
32.403 8.3968 5.5558 3.5395 2.4402 1.9537
33.129 8.6098 5.7968 3.7123 2.4759 2.0409
34.399 9.0073 6.2965 4.0336 2.5166 2.028
35.669 9.2383 6.7298 4.3787 2.607 2.0385
36.395 10.721 6.8432 4.5446 2.698 2.2459
39.479 13.412 7.3227 5.2796 3.023 2.3496
40.749 8.6375 7.7677 5.6155 3.1161 2.4002
41.475 9.2485 13.345 5.2701 3.2681 2.4571
43.289 5.3859 11.602 5.4409 3.4952 2.6021
44.559 4.1633 9.4111 5.817 3.6986 2.6408
45.829 3.6298 6.8483 5.6632 3.9345 2.6414
47.099 2.7678 5.742 6.2548 3.9294 2.7945
48.369 2.5314 4.4132 5.519 4.064 2.8262

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49.639 2.0387 3.815 4.8841 4.1929 2.9251
 50.909 1.8894 3.1747 4.1704 4.2237 2.8757
 52.179 1.6542 2.2633 3.0053 4.4283 2.9868
 53.449 1.2176 2.3245 3.1682 3.8959 3.3574
 54.719 1.3379 2.5095 3.0227 4.5734 2.9917
 55.989 1.9329 1.645 2.2605 4.6809 3.1566
 57.259 1.0041 1.4239 1.9605 4.5097
 58.529 1.0719 4.2253

SKIN FRICTION DISTRIBUTIONS AT MACH 3

CENTERLINE SKIN FRICTION DISTRIBUTIONS

7X7 DEGREE INTERACTION

Z/60	CF	EE+03	ERROR+03
11.611	1.41	0.141	
15.24	1.436	0.1436	
18.869	1.303	0.1303	
22.497	1.196	0.1196	
26.126	1.043	0.1043	
29.754	0.937	0.0937	
33.383	0.858	0.0858	
37.011	0.762	0.0762	
40.64	0.734	0.0734	
44.269	0.713	0.0713	

11x11 DEGREE INTERACTION

Z/60	CF	EE+03	ERROR+03
7.9829	1.462	0.1462	
11.611	1.407	0.1407	
15.24	1.173	0.1173	
18.869	0.594	0.0594	
22.272	0.336	0.03452	
26.126	0.135	0.135	
29.754	0.13	0.13	
33.383	0.631	0.0631	
37.011	1.831	0.1831	
40.64	1.883	0.1883	

SPANWISE SKIN FRICTION DISTRIBUTIONS

7X7 DEGREE INTERACTION @ Z/60=40.6

X/60	CF	EE+03	ERROR+03
0	0.734	0.0734	
1.8143	1.227	0.1227	
2.7214	1.733	0.1733	
3.6286	2.35	0.235	
4.5429	2.63	0.263	
5.45	3.005	0.3005	
6.35	3.04	0.4	
7.2571	2.34	0.234	
7.9393			

11X11 DEGREE INTERACTION

Z/60	CF	EE+03	ERROR+03
0	0.13	0.03078	
0.9071	1.364	0.1364	
2.7214	2.966	0.2966	

4.5357 3.828 0.3828 0.3897
 5.4429 3.79 0.379
 6.4661 3.744 0.3744

SKIN FRICTION DISTRIBUTIONS FOR MACH 4, 15x15 DEGREE INTERACTION

CENTERLINE SKIN FRICTION DISTRIBUTION

Z/60 CF EE+03 ERROR+03

12.417	1.099	0.1099	
16.495	0.564	0.0564	
18.564	0.304	0.05	
23.753	0	0.1	
25.567	1.022	0.183	
27.381	1.918	0.1918	
31.01	1.845	0.1845	
34.638	1.877	0.1877	
38.267	3.116	0.3116	

SPANWISE SKIN FRICTION DISTRIBUTION

Z/60 = 25.3

X/60 CF EE+03 ERROR+03

0	1.022	0.1022	
0.90714	0.582	0.06028	
2.0973	2.368	0.2368	
2.6053	1.053	0.1053	
3.9697	2.905	0.2905	
5.1598	4.049	0.4049	
6.125	5.344	0.5344	
6.5314	4.552	0.4552	
7.2571	3.931	0.3931	

FLORFIELD PITOT SURVEY DATA

MACH 4, 15x15 DEGREE INTERACTION

SPANWISE PLANE "C" LOCATED AT Z/60 = 32.33

X mm	Y mm	PtZ/p0
15.270480	2.374900	2.38831E-01
15.262860	3.276600	2.325127E-01
15.260320	4.295140	2.599282E-01
15.252700	5.300980	2.540075E-01
15.252700	6.286500	2.372937E-01
15.245080	7.207420	2.744129E-01
15.247620	8.336280	2.92614E-01
15.240000	9.362400	2.909256E-01
15.247620	10.370820	2.919231E-01
15.252700	11.361420	2.777345E-01
15.245080	12.400280	2.742993E-01
15.245080	13.413740	2.915159E-01
15.242540	14.417040	2.960328E-01
15.245080	15.394940	2.921575E-01
15.247620	16.375300	2.882945E-01
15.247620	17.330420	2.801678E-01
15.250160	18.321020	2.744834E-01
15.247620	19.372580	2.869852E-01
15.255240	20.474940	3.03747E-01
15.252700	21.531580	2.963652E-01
15.247620	22.532340	3.063171E-01
15.252700	23.515320	3.039299E-01
15.247620	24.490680	3.075611E-01
15.247620	25.466360	2.924217E-01

NOTE: THE EXCERPT
 SHOWN HERE IS FOR
 SAMPLE PURPOSES ONLY.
 SPACE LIMITATIONS
 PRECLUDE A COMPLETE
 LISTING.

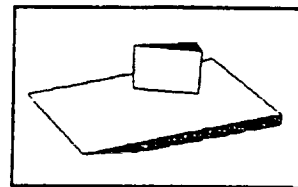
Ref.: 12-13

Author: Hsu and Settles

Geometry: 3-D Fin

Mach number: 3, 4

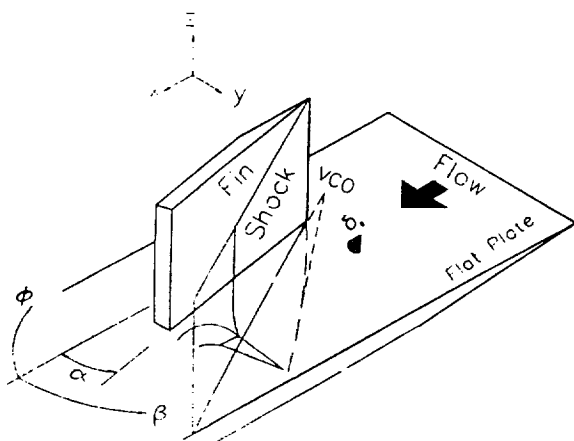
Data: flowfield density maps



The experiments were carried out using a single sharp unswept fin mounted at angle-of-attack on a flat plate in the supersonic wind tunnel facility of the Penn State Gas Dynamics Laboratory. The flat-plate boundary-layers at Mach 3 and 4 are the same as those previously documented in Ref. 1 (KIM.DAT). These are equilibrium turbulent boundary-layers developing naturally on a flat plate at high Reynolds number, and are essentially adiabatic.

The swept, single-fin shock/boundary-layer interaction was studied by way of conical holographic interferometry, wherein the holographic test beam was focused at or near the virtual origin of the quasiconical swept interaction. The light rays then coincided with the rays of the interaction, rendering the interaction two-dimensional insofar as the quasiconical approximation is valid. Users of these data who are unfamiliar with this approach should first consult the cited references and Alvi and Settles, *AIAA Journal*, Vol. 30, Sept. 1992, pp. 2252-2258.

Holographic interferograms obtained as described above and in the cited references were reduced to provide flowfield density data, assuming that each interaction was a two-dimensional flow in the angular coordinates β and ϕ defined in the diagram below. β is the azimuthal coordinate measured from the freestream direction, and is positive in the direction illustrated. ϕ is the elevation coordinate measured from the plane of the flat plate, and is also positive in the direction illustrated. Both β and ϕ have vertices at the virtual conical origin (VCO) of the interaction, which is located on the flat plate ahead of the fin leading edge, along a line extrapolated from the inviscid shock angle, β_0 . The distance from the fin leading edge to the VCO for the three interactions studied here were: 48 ± 6 mm for the Mach 3, $\alpha = 10^\circ$ case, 33 ± 3 mm for the Mach 3, $\alpha = 16^\circ$ case, and 21 ± 3 mm for the Mach 4, $\alpha = 20^\circ$ case.



The data file HSU.DAT contains preliminary information and density data files for each of these 3 interactions. The extensive flowfield density data are given in columnar form, the number of β columns ranging from 18 to 36 and the number of ϕ rows ranging from 58 to 100, depending on the overall angular extent of the interaction in question. At each flowfield location defined by a β, ϕ pair, the measured static density is given along with its normalization by the freestream static density.

Density data from conical holographic interferogram
of swept shock wave/turbulent 8. L. interaction

Mach 3.0 Fin Angle 10.1 degrees

Penn State Gas Dynamics Laboratory, March 1991

Density data from conical holographic interferogram
of swept shock wave/turbulent 8. L. interaction

Mach 3.0 Angle 16.0 degrees

Penn State Gas Dynamics Laboratory, March 1991

Density data from conical holographic interferogram
of swept shock wave/turbulent 8. L. interaction

Mach 4 Angle 20 degrees

Penn State Gas Dynamics Laboratory, March 1991

INCOMING BOUNDARY LAYER PARAMETERS AT X=178 mm FROM FLAT-PLATE
LEADING EDGE

Mach 3 (actual freestream Mach number was 2.91 for this survey)

Delta = 3.02 mm

Delta* = 0.895 mm

Theta = 0.184 mm

Cf = 0.001520

Re(theta) = 9751

Case: m3a10/m3a10a

Beta Column Count= 18

Phi Row Count= 58

Freestream Mach Number= 2.95

Angle of Attack= 10.1

Fin L.E. from flat plate L.E., X= 10.414 cm

Fin L.E. from side wall, Y= 3.333750 cm

Freestream stagnation Temperature.= 296.3000 K

Freestream stagnation Pressure= 0.6892000 MPa

Freestream Density RhoInf = 0.6189000 Kg/m³

Reynolds number 6.19E+07/m

Tu/Taw = 1.06 (roughly adiabatic)

Col.#	Beta	Phi	Rho	Rho/RhoInf
1	34.473	9.235	0.621	1.00
1	34.473	9.198	0.621	1.00
1	34.473	9.161	0.621	1.00
1	34.473	9.124	0.621	1.00
1	34.473	9.087	0.621	1.00
1	34.473	9.051	0.621	1.00
1	34.473	9.014	0.621	1.00
1	34.473	8.977	0.621	1.00
1	34.473	8.940	0.621	1.00
1	34.473	8.903	0.621	1.00
1	34.473	8.866	0.621	1.00
1	34.473	8.830	0.621	1.00
1	34.473	8.793	0.621	1.00
1	34.473	8.756	0.621	1.00
1	34.473	8.719	0.621	1.00
1	34.473	8.682	0.621	1.00
1	34.473	8.646	0.621	1.00
1	34.473	8.609	0.621	1.00
1	34.473	8.572	0.621	1.00
1	34.473	8.535	0.621	1.00
1	34.473	8.498	0.621	1.00
1	34.473	8.462	0.621	1.00
1	34.472	8.277	0.623	1.01
1	34.475	8.080	0.623	1.01

EXCERPT ONLY
SEE HSU.DAT

INCOMING BOUNDARY LAYER PARAMETERS AT X=178 mm FROM FLAT-PLATE
LEADING EDGE

Mach 3 (actual freestream Mach number was 2.91 for this survey)

Delta = 3.02 mm

Delta* = 0.895 mm

Theta = 0.184 mm

Cf = 0.001520

Re(theta) = 9751

Case: m3a16/m3a16

Beta Column Count= 24

Phi Row Count= 92

Freestream Mach Number= 2.950000

Angle of Attack= 16.00000

Fin L.E. from flat plate L.E., X= 11.68400 cm

Fin L.E. from side wall, Y= 2.698750 cm

Freestream stagnation Temperature.= 289.0000 K

Freestream stagnation Pressure= 0.7565000 MPa

Freestream Density RhoInf = 0.6238000 Kg/m³

Reynolds number 6.19E+07/m

Tu/Taw = 1.06 (roughly adiabatic)

Col.#	Beta	Phi	Rho	Rho/RhoInf
1	46.111	14.523	0.623	1.00
1	46.111	14.520	0.623	1.00
1	46.111	14.517	0.623	1.00
1	46.111	14.514	0.623	1.00
1	46.111	14.511	0.623	1.00
1	46.111	14.508	0.623	1.00
1	46.111	14.505	0.623	1.00
1	46.111	14.502	0.623	1.00
1	46.111	14.499	0.623	1.00
1	46.111	14.496	0.623	1.00
1	46.111	14.493	0.623	1.00
1	46.111	14.490	0.623	1.00
1	46.111	14.487	0.623	1.00
1	46.111	14.484	0.623	1.00
1	46.111	14.480	0.623	1.00
1	46.111	14.477	0.623	1.00
1	46.111	14.474	0.623	1.00
1	46.111	14.471	0.623	1.00
1	46.111	14.468	0.623	1.00
1	46.111	14.465	0.623	1.00
1	46.111	14.462	0.623	1.00
1	46.111	14.459	0.623	1.00
1	46.111	14.456	0.623	1.00
1	46.111	14.453	0.623	1.00
1	46.111	14.450	0.623	1.00

EXCERPT ONLY
SEE HSU.DAT

INCOMING BOUNDARY LAYER PARAMETERS AT X=178 mm FROM FLAT-PLATE
LEADING EDGE

Mach 4 (actual freestream Mach number was 3.88 for this survey)

Delta = 2.67 mm

Delta* = 0.950 mm

Theta = 0.128 mm

Cf = 0.001325

Re(theta) = 9082

Case: m4a20/m4a20

Beta Column Count= 36

Phi Row Count= 100

Freestream Mach Number= 3.930000

Angle of Attack= 20.00000

Fin L.E. from flat plate L.E., X= 10.00125 cm

Fin L.E. from side wall, Y= 2.460625 cm

Wd from fin leading edge= 17 mm (along the inviscid shock)

Freestream stagnation Temperature.= 294.8000 K

Freestream stagnation Pressure= 1.385840 MPa

Freestream Density RhoInf = 0.4841200 Kg/m³

Reynolds number 6.79E+07/m

Tu/Taw = 1.06 (roughly adiabatic)

Col.#	Beta	Phi	Rho	Rho/RhoInf
1	49.284	13.244	0.485	1.00
1	49.284	13.238	0.485	1.00
1	49.284	13.233	0.485	1.00
1	49.284	13.227	0.485	1.00
1	49.284	13.222	0.485	1.00
1	49.284	13.216	0.485	1.00
1	49.284	13.211	0.485	1.00
1	49.284	13.205	0.485	1.00
1	49.284	13.200	0.485	1.00
1	49.284	13.195	0.485	1.00
1	49.284	13.189	0.485	1.00
1	49.284	13.184	0.485	1.00
1	49.284	13.178	0.485	1.00
1	49.284	13.173	0.485	1.00
1	49.284	13.167	0.485	1.00
1	49.284	13.162	0.485	1.00
1	49.284	13.156	0.485	1.00
1	49.284	13.151	0.485	1.00
1	49.284	13.145	0.485	1.00
1	49.284	13.140	0.485	1.00
1	49.284	13.134	0.485	1.00
1	49.284	13.129	0.485	1.00
1	49.284	13.124	0.485	1.00

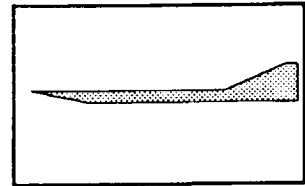
EXCERPT ONLY
SEE HSU.DAT

Author: Kuntz, D. W., *et al*

Geometry: 2-D Compression Corner

Mach Number: 3

Data: p_{wall} , mean & fluctuating flowfield surveys (two channel
Laser Doppler Velocimetry)



Kuntz, D. W., Amatucci, V. A., and Addy, A. L., "An Experimental Study of the Shock Wave-Turbulent Boundary Layer Interaction," Presented at the Winter Annual Meeting of the American Society of Mechanical Engineers, Miami Beach, Florida, November 17-22, 1985, and published in the International Symposium on Laser Anemometry, FED-Vol. 33, ed. A. Dybbs and P. A. Pfund, pp. 173-178.

Kuntz, D. W., Amatucci, V. A., and Addy, A. L., "Turbulent Boundary-Layer Properties Downstream of the Shock-Wave/Boundary-Layer Interaction," *AIAA Journal*, Vol. 25, No. 5, pp. 668-675, May, 1987.

Kuntz, D. W., "An Experimental Investigation of the Shock Wave-Turbulent Boundary Layer Interaction," Ph.D. Thesis, Dept. of Mechanical and Industrial Engineering, Univ. of Illinois at Urbana-Champaign, Urbana, Illinois, 1985.

Experiments were conducted in a supersonic wind tunnel with a 10.2 x 10.2 cm test section. The models were two-dimensional compression corners which spanned the test section. Experiments were conducted with corner angles of 8, 12, 16, 20, and 24°.

The freestream Mach number, determined from surface pressure measurements, was approximately 2.94, and the stagnation pressure was maintained at approximately 483 kPa (70 psia) for all data points.

The primary measurement tool used in this investigation was a two-color laser Doppler velocimeter system. This system was used to obtain two-component mean velocity and turbulent property measurements in both the upstream and redeveloping downstream boundary layers within the flowfields. In addition to the LDV measurements, surface static pressure measurements, surface streak pattern measurements, and high-speed Schlieren photographs were also taken.

Measurements made in the wind tunnel in the absence of the compression corner models indicated that the undisturbed turbulent boundary layer had a thickness of 8.27 mm ($u_e = 0.99 u_\infty$), a displacement thickness of 3.11 mm and a momentum thickness of 0.57 mm.

The tabular data includes the surface static pressure distribution, the LDV data from a single traverse made upstream of the interaction, and the LDV data from the traverses made downstream of the interaction. Each traverse made downstream of the interaction includes data taken at a single point above the shock wave. The coordinate systems used are presented in the accompanying figure, and the definitions of the quantities listed are included in the accompanying table. The reader is referred to the *AIAA Journal* article cited above for a detailed discussion of the corrections applied to the LDV data, the errors associated with the LDV data, and estimates of the measurement accuracies.

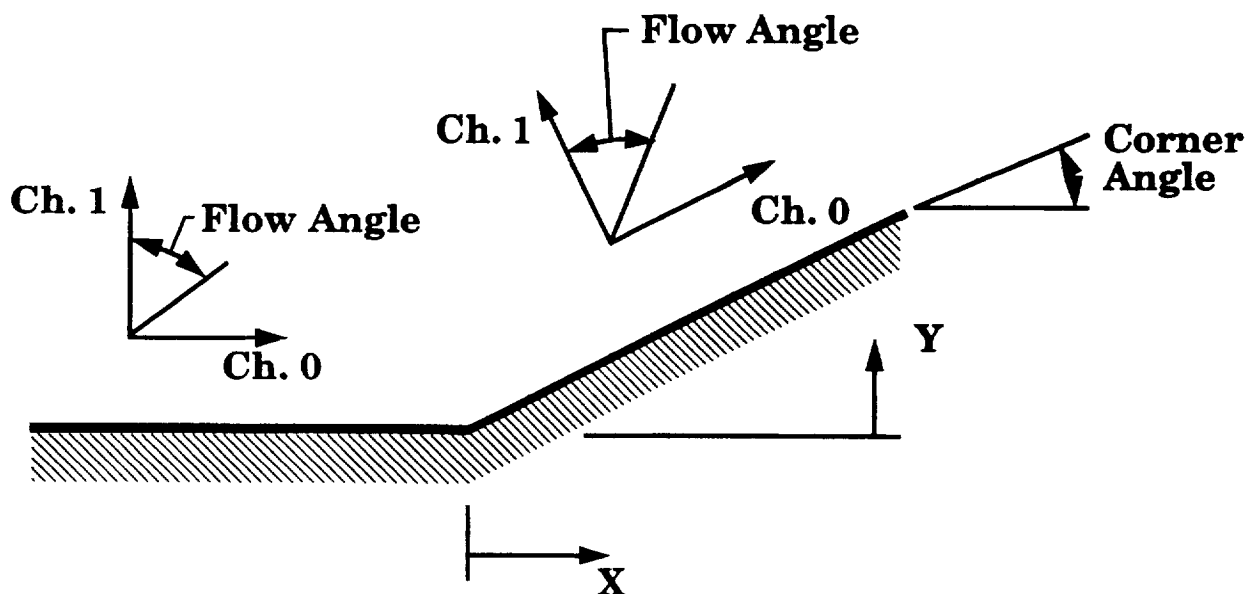
Definitions of the Tabular LDV Data

<u>Variable Name</u>	<u>Description</u>	<u>Units</u>	<u>Definition</u>	<u>Note</u>
x	Longitudinal Position	mm	See Figure	
y	Vertical Position	mm	See Figure	
Vt	Total Velocity	m/s	$\sqrt{(V0)^2 + (V1)^2}$	1
theta	Flow Angle	deg	See Figure	
V0	Average Velocity, Ch 0	m/s	See Figure	
V1	Average Velocity, Ch 1	m/s	See Figure	
var0	Variance, Ch 0	m ² /s ²	$\frac{\sum (v0 - V0)^2}{n - 1}$	2
var1	Variance, Ch 1	m ² /s ²	$\frac{\sum (v1 - V1)^2}{n - 1}$	
s01	Reynolds Stress Term	m ² /s ²	$\overline{(v0 - V0)(v1 - V1)}$	3
s001	Triple Product	m ³ /s ³	$\overline{(v0 - V0)^2(v1 - V1)}$	
s110	Triple Product	m ³ /s ³	$\overline{(v0 - V0)(v1 - V1)^2}$	
s000	Triple Product	m ³ /s ³	$\overline{(v0 - V0)^3}$	
s111	Triple Product	m ³ /s ³	$\overline{(v1 - V1)^3}$	
sk0	Skewness, Ch 0	-	$(s000) / (\sqrt{\text{var0}})^3$	

<u>Variable Name</u>	<u>Description</u>	<u>Units</u>	<u>Definition</u>	<u>Note</u>
sk1	Skewness, Ch 1	-	$(s111) / (\sqrt{\text{var1}})^3$	
ku0	Flatness, Ch 0	-	$[(s0000) / (\sqrt{\text{var0}})^4] - 3.0$	4
ku1	Flatness, Ch 1	-	$[(s1111) / (\sqrt{\text{var1}})^4] - 3.0$	

Notes:

- 1 V denotes average velocity, v denotes instantaneous velocity. Upstream of the compression corner, Channel 0 is parallel to the tunnel floor and Channel 1 is perpendicular to the tunnel floor. Downstream of the compression corner, Channel 0 is parallel to the ramp surface, and Channel 1 is perpendicular to the ramp surface.
- 2 The variance is the square of the standard deviation. The turbulence intensity would be found by taking the square root of the variance and dividing by the appropriate reference quantity (such as a reference velocity or a local velocity).
- 3 The overbar denotes a simple average of the instantaneous measurements.
- 4 The quantity s0000 was calculated in a similar manner as s000.



Coordinate System

Kuntz et al 2-D Compression Corner Data

20. Degree Compression Corner

M-inf = 2.94
p0 = 483 kPa
t0 = 30 deg C
delta-0 = 8.27 mm

Separation Location = -1.63*delta-0
Reattachment Location = 0.52*delta-0 (measured parallel to the ramp surface)

Mean Surface Static Pressure Distribution

x (mm)	p/p-inf
-71.12	1.000
-66.04	0.992
-60.96	0.986
-55.88	0.996
-50.80	0.996
-48.26	0.998
-45.72	0.995
-43.18	0.992
-40.64	0.999
-38.10	1.008
-35.56	0.999
-33.02	1.009
-30.48	0.994
-27.94	1.007
-25.40	0.991
-22.86	0.992
-20.32	1.000
-17.78	1.042
-15.24	1.168
-12.70	1.329
-10.16	1.677
-7.62	1.817
-5.08	1.899
-2.54	1.970
0.36	2.060
2.39	2.192
4.77	2.356
7.16	2.562
9.55	2.692
11.93	2.792
14.32	2.951
16.71	2.995
19.09	3.096
21.48	3.185
23.87	3.225
26.26	3.277
28.64	3.352
31.03	3.392
33.42	3.437
35.80	3.468
38.19	3.496
40.58	3.487
42.96	3.480

NOTE: DUE TO SPACE
LIMITATIONS, ONLY THE 20° DATA
ARE TABULATED HERE. SEE
DISKETTE FILE KUNTZ.DAT FOR
COMPLETE TABULATIONS FOR
ALL FIVE COMPRESSION
CORNERS.

Laser Doppler Velocimeter Data

Basic data, x = -20.0 mm.

y	Vt	theta	VO	Vl	var0	var1	s01
15.00	629.83	90.02	629.83	-0.24	123.974	67.884	-22.16
14.00	631.15	89.97	631.15	0.31	82.210	74.888	1.03
13.00	631.04	90.01	631.04	-0.06	83.261	66.507	-10.52
12.00	631.18	89.97	631.18	0.28	82.380	63.546	-6.84
11.00	630.37	89.98	630.37	0.25	91.020	64.016	-3.13
10.00	628.97	89.97	628.97	0.38	144.848	64.641	-14.42
9.50	628.92	90.02	628.92	-0.18	139.109	93.333	11.20
9.00	625.96	90.05	625.96	-0.54	179.863	89.770	10.54
8.50	624.47	90.02	624.47	-0.23	224.517	94.184	-0.82
8.00	623.16	90.05	623.16	-0.55	268.612	109.647	-12.23
7.50	621.40	89.94	621.40	0.65	279.304	128.847	-31.07
7.00	616.24	89.99	616.24	0.14	330.528	130.878	-18.25
6.50	609.41	89.90	609.40	1.02	501.894	148.948	-102.43
6.00	601.19	90.03	601.19	-0.27	592.226	181.281	-67.81
5.50	588.01	89.96	588.01	0.39	798.860	191.253	-120.55
5.00	580.70	90.00	580.70	0.04	931.395	195.745	-163.73
4.50	569.92	90.08	569.92	-0.80	945.322	235.213	-159.33
4.00	555.81	90.12	555.81	-1.20	1274.509	236.399	-211.54
3.50	544.79	90.25	544.78	-2.38	1255.189	245.164	-219.28
3.00	528.84	90.21	528.84	-1.94	1530.370	278.494	-270.16
2.50	510.50	90.21	510.50	-1.86	1616.059	239.195	-228.87
2.25	502.73	90.28	502.73	-2.46	1738.444	252.180	-283.66
2.00	490.31	90.28	490.30	-2.41	1962.803	235.178	-324.92
1.75	482.27	90.39	482.26	-3.29	1865.482	237.170	-291.96
1.50	468.59	90.35	468.59	-2.83	1853.864	225.861	-283.13

Higher order moment data, x = -20.0 mm.

y	s001	s110	s000	s111	sk0	sk1	ku0	ku1
15.00	2.0235E+02	-9.3537E+01	-2.2967E+03	6.8859E+01	-1.6639E+00	1.2311E-01	5.4483E+00	4.2745E-01
14.00	-5.0543E+02	-3.6605E+02	-1.0831E+03	-2.9269E+02	-1.4531E+00	-4.5163E-01	8.0608E+00	2.1114E+00
13.00	1.5715E+02	-4.0176E+01	-1.1367E+03	7.3030E+01	-1.4961E+00	1.3465E-01	8.9324E+00	2.7036E-01
12.00	2.3253E+01	3.8892E+01	-7.4730E+02	4.5761E+01	-9.9945E-01	9.0336E-02	5.9122E+00	3.4205E-01
11.00	-2.3082E+02	-2.2224E+01	-1.6080E+03	-3.4294E+01	-1.8517E+00	-6.6954E-02	9.2173E+00	4.2448E-01
10.00	1.7118E+02	-2.1015E+01	-3.3750E+03	-1.6216E+01	-1.9360E+00	-3.1202E-02	7.7342E+00	1.9883E-01
9.50	-1.2724E+03	-1.6979E+03	-3.2023E+03	-1.4625E+03	-1.9518E+00	-1.6220E+00	7.4695E+00	1.2167E+01
9.00	-9.8147E+02	-9.4948E+02	-4.4037E+03	-8.4440E+02	-1.8256E+00	-9.9278E-01	5.6903E+00	6.4585E+00
8.50	-8.2099E+02	-1.1794E+03	-6.3828E+03	-6.7499E+02	-1.8973E+00	-7.3847E-01	5.6451E+00	5.6307E+00
8.00	-6.2678E+02	-1.7881E+03	-5.4699E+03	-6.2733E+02	-1.2425E+00	-5.4639E-01	2.0279E+00	7.0365E+00
7.50	-6.0394E+02	-2.3652E+03	-6.9227E+03	-1.5742E+03	-1.4831E+00	-1.0764E+00	3.3418E+00	7.2633E+00
7.00	-1.5415E+03	-2.5645E+03	-7.9463E+03	-1.1649E+03	-1.3224E+00	-7.7804E-01	2.9939E+00	5.5822E+00
6.50	1.6567E+03	-2.2157E+03	-1.2144E+04	-4.1840E+02	-1.0800E+00	-2.3016E-01	1.3151E+00	3.9066E+00

6.00	-7.5540E+01	-3.6699E+03	-1.1941E+04	-2.6558E+03	-8.2853E-01	-1.0881E+00	5.5174E-01	7.2079E+00
5.50	-1.5719E+03	-3.3634E+03	-1.6858E+04	-7.7005E+02	-7.4663E-01	-2.9114E-01	5.9426E-01	2.8110E+00
5.00	-1.7549E+03	-2.5210E+03	-1.5825E+04	-1.5825E+02	-5.5672E-01	-6.0995E-02	9.4438E-03	2.1575E+00
4.50	-6.9012E+02	-3.8577E+03	-1.5033E+04	-9.2446E+02	-5.1723E-01	-2.5627E-01	3.4463E-01	2.5078E+00
4.00	-6.0964E+02	-2.1747E+03	-1.8942E+04	4.4923E+02	-4.1631E-01	1.2359E-01	1.1514E-01	1.3522E+00
3.50	-1.0611E+03	-2.0295E+03	-1.6478E+04	3.8577E+02	-3.7056E-01	1.0049E-01	6.4845E-02	1.0624E+00
3.00	-1.1746E+03	-2.0313E+03	-1.3114E+04	-6.7744E+00	-2.1905E-01	-1.4576E-03	-2.6129E-01	2.1297E+00
2.50	8.3028E+02	-5.5184E+01	-7.5547E+03	9.3189E+00	-1.1629E-01	2.5191E-03	-1.5630E-01	1.0117E+00
2.25	-1.6478E+03	-5.1679E+02	-1.3447E+04	-6.5488E+02	-1.8552E-01	-1.6353E-01	-2.4052E-01	1.3350E+00
2.00	-2.3778E+03	-5.0284E+02	-1.1720E+04	1.4881E+02	-1.3477E-01	4.1262E-02	-2.9848E-01	3.8375E-01
1.75	-1.6490E+03	-1.0535E+03	1.9196E+03	-2.3046E+02	2.3825E-02	-6.3098E-02	-2.1967E-01	1.1392E+00
1.50	-2.3348E+03	-1.3341E+03	-9.4191E+03	1.1719E+03	-1.1800E-01	3.4524E-01	-1.9256E-01	3.1097E+00

Basic data, x = 15.0 mm.

y	Vt	theta	VO	VI	var0	var1	s01
25.00	626.64	110.04	588.69	-214.76	89.488	83.653	-40.67
15.00	590.13	103.08	574.82	-133.58	138.698	1179.915	-41.62
14.50	584.99	101.68	572.88	-118.42	143.080	1220.481	-85.82
14.00	579.89	100.38	570.40	-104.46	346.483	1524.398	-351.28
13.50	573.35	99.45	565.56	-94.17	702.691	1821.974	-637.51
13.00	562.01	98.48	555.86	-82.87	1556.716	2070.953	-1091.02
12.50	534.57	97.12	530.45	-66.26	4221.431	2317.852	-2184.88
12.00	523.41	97.25	519.22	-66.07	5794.362	2657.328	-2846.59
11.50	457.37	94.88	455.72	-38.88	9134.602	2521.922	-3574.67
11.00	419.16	93.62	418.32	-26.43	10370.281	2475.441	-3922.39
10.50	389.49	93.14	388.91	-21.33	11522.217	2797.428	-4310.29
10.00	364.97	93.54	364.28	-22.54	12994.065	2594.220	-4163.35
9.50	354.50	94.25	353.52	-26.30	11880.019	2700.931	-4253.63
9.00	331.57	94.79	330.41	-27.71	12764.326	2801.589	-4324.87
8.50	291.62	93.95	290.93	-20.10	14428.071	3244.977	-4959.70
8.00	231.80	93.48	231.37	-14.05	11345.315	2985.999	-4142.82
7.75	217.11	93.14	216.79	-11.87	10872.860	3098.062	-4195.36
7.50	207.76	92.54	207.55	-9.22	10031.171	2955.841	-3868.81
7.25	201.24	92.78	201.01	-9.76	9021.621	2923.591	-3668.41
7.00	179.28	91.23	179.24	-3.86	7698.342	2660.897	-3053.32

Higher order moment data, x = 15.0 mm.

y	s001	s110	s000	s111	sk0	sk1	ku0	ku1
25.00	3.8656E+02	-2.3489E+02	-7.6392E+02	1.4088E+02	-9.0240E-01	1.8413E-01	2.9275E+00	1.1219E-01
15.00	-2.6332E+02	-5.4143E+03	-2.3439E+03	-1.3941E+04	-1.4349E+00	-3.4398E-01	7.8909E+00	1.5692E-01
14.50	-5.3550E+02	-3.5512E+03	-1.9059E+03	-1.7902E+04	-1.1136E+00	-4.1986E-01	4.1317E+00	2.5554E-01
14.00	2.9727E+04	-2.7298E+04	-3.3843E+04	-5.9964E+03	-5.2474E+00	-1.0075E-01	4.7258E+01	1.7821E+00
13.50	7.0572E+04	-5.4221E+04	-9.3617E+04	1.1285E+04	-5.0258E+00	1.4510E-01	3.5415E+01	1.8272E+00
13.00	1.1299E+05	-6.2639E+04	-2.0635E+05	6.5404E+02	-3.3593E+00	6.9399E-03	1.4209E+01	9.2126E-01
12.50	2.3912E+05	-9.7709E+04	-6.1925E+05	5.4779E+03	-2.2577E+00	4.9089E-02	5.4012E+00	7.8636E-01
12.00	3.6922E+05	-1.3696E+05	-9.8090E+05	8.2362E+03	-2.2239E+00	6.0125E-02	5.4016E+00	8.2041E-01
11.50	2.2138E+05	-7.2436E+04	-8.1353E+05	6.2758E+03	-9.3183E-01	4.9553E-02	3.0818E-01	9.2357E-01
11.00	2.1316E+05	-7.5393E+04	-8.0677E+05	5.4406E+04	-7.6395E-01	4.4174E-01	2.4558E-01	5.4881E-01
10.50	2.0007E+05	-9.9943E+04	-7.9622E+05	7.0847E+04	-6.4377E-01	4.7883E-01	-3.8348E-02	5.2800E-01
10.00	1.5987E+05	-7.1827E+04	-1.0101E+06	6.2538E+04	-6.8196E-01	4.7330E-01	1.8900E-01	4.3730E-01
9.50	1.6260E+05	-8.1292E+04	-6.2072E+05	5.7436E+04	-4.7937E-01	4.0918E-01	-3.0599E-01	3.7710E-01
9.00	1.5333E+05	-8.8085E+04	-6.9643E+05	7.3656E+04	-4.8292E-01	4.9671E-01	-3.0711E-01	4.5722E-01
8.50	3.7100E+04	-4.4120E+04	-3.7368E+05	5.6335E+04	-2.1562E-01	3.0476E-01	-6.8503E-01	2.5830E-02

8.00	-1.2782E+05	2.4610E+04	3.8752E+05	1.4066E+04	3.2068E-01	8.6208E-02	-4.7752E-01	-1.7768E-01
7.75	-1.8393E+05	5.8723E+04	4.6173E+05	-4.9383E+03	4.0726E-01	-2.8638E-02	-3.3945E-01	-1.9150E-01
7.50	-1.9670E+05	6.7295E+04	4.4413E+05	-8.5046E+03	4.4206E-01	-5.2922E-02	-2.1879E-01	-2.7334E-01
7.25	-1.9442E+05	7.3813E+04	4.9625E+05	-1.7954E+04	5.7912E-01	-1.1357E-01	-5.5344E-02	-2.9367E-01
7.00	-1.6831E+05	7.2865E+04	3.8035E+05	-2.4550E+04	5.6311E-01	-1.7886E-01	1.2360E-01	-2.1235E-01

Basic data, x = 20.0 mm.

y	Vt	theta	VO	VI	var0	var1	s01
30.00	625.05	110.03	587.25	-214.08	136.945	126.725	-77.50
17.50	582.26	100.97	571.61	-110.81	119.179	1047.679	-84.41
17.00	579.03	99.93	570.36	-99.82	118.294	1259.272	-135.87
16.50	576.45	99.82	568.01	-98.29	189.085	1855.030	-130.32
16.00	571.51	97.78	566.25	-77.36	159.283	1544.215	-190.19
15.50	566.68	97.12	562.32	-70.22	558.570	1451.376	-432.08
15.00	560.39	97.06	556.14	-68.84	978.371	1923.282	-775.45
14.50	546.04	96.17	542.88	-58.68	2660.936	2009.862	-1433.50
14.00	526.37	95.67	523.79	-52.03	5104.213	2545.022	-2422.73
13.50	516.33	95.35	514.08	-48.15	4362.667	1901.045	-1910.74
13.00	440.56	93.86	439.56	-29.67	9047.953	1922.597	-2681.29
12.00	410.01	94.06	408.99	-29.00	10749.297	2177.956	-3260.18
11.50	393.82	94.34	392.69	-29.80	9239.213	2076.666	-2929.18
11.00	378.32	94.12	377.34	-27.20	8944.463	2288.416	-3042.51
10.50	335.24	93.23	334.71	-18.90	11318.126	2961.173	-4106.47
10.00	308.70	92.79	308.34	-15.04	11402.291	3121.934	-4045.75
9.75	263.09	92.83	262.77	-13.00	10810.626	3038.149	-3913.95
9.50	244.09	91.86	243.96	-7.92	10332.898	2927.825	-3753.18
9.25	238.82	91.89	238.69	-7.87	9913.989	3076.881	-3762.51
9.00	222.19	91.21	222.14	-4.71	8949.622	2750.721	-3424.42
8.75	209.10	90.03	209.10	-0.12	7419.993	2305.558	-2680.20

Higher order moment data, x = 20.0 mm.

y	s001	s110	s000	s111	sk0	sk1	ku0	ku1
30.00	1.9534E+03	-1.8259E+03	-2.5761E+03	1.7093E+03	-1.6074E+00	1.1982E+00	6.0317E+00	5.5374E+00
17.50	-9.8253E+02	-7.9655E+02	-1.5069E+03	-3.1088E+04	-1.1582E+00	-9.1674E-01	4.8702E+00	9.2906E-01
17.00	-7.4653E+02	-4.9095E+02	-3.5741E+01	-4.4977E+04	-2.7779E-02	-1.0065E+00	2.8440E+00	1.2101E+00
16.50	-4.3219E+02	-2.2998E+02	-2.6020E+03	-7.1754E+04	-1.0007E+00	-8.9809E-01	5.3332E+00	4.5549E-01
16.00	2.1981E+03	-2.5348E+03	-2.0406E+03	-6.7117E+04	-1.0153E+00	-1.1060E+00	2.8211E+00	2.2282E+00
15.50	3.8359E+04	-2.8690E+04	-5.7306E+04	-2.7934E+04	-4.3409E+00	-5.0522E-01	2.9728E+01	3.0239E+00
15.00	8.0443E+04	-6.6852E+04	-1.2359E+05	-4.1177E+04	-4.0325E+00	-4.8748E-01	2.1020E+01	2.2281E+00
14.50	1.7635E+05	-7.4881E+04	-3.9775E+05	-2.6500E+04	-2.8978E+00	-2.9410E-01	9.0649E+00	1.8928E+00
14.00	3.4152E+05	-1.5396E+05	-8.9468E+05	7.3953E+03	-2.4534E+00	-7.5997E-02	6.3587E+00	1.9723E+00
13.50	1.6840E+05	-5.7302E+04	-4.8785E+05	-1.2573E+04	-1.6930E+00	-1.5169E-01	2.7021E+00	1.7742E+00
13.00	2.0158E+05	-8.0167E+04	-8.5224E+05	3.2897E+04	-9.9023E-01	3.9023E-01	7.8096E-01	1.8299E+00
12.00	1.7734E+05	-6.8434E+04	-7.0653E+05	3.4743E+04	-6.3396E-01	3.4181E-01	-2.8864E-01	8.0596E-01
11.50	1.3043E+05	-6.9456E+04	-6.0370E+05	4.6776E+04	-6.7978E-01	4.9427E-01	1.9807E-01	5.4240E-01
11.00	1.5922E+05	-8.6112E+04	-6.0859E+05	6.2820E+04	-7.1943E-01	5.7385E-01	6.6930E-01	7.9868E-01
10.50	1.5846E+05	-9.8766E+04	-5.3539E+05	8.5750E+04	-4.4464E-01	5.3215E-01	-3.2403E-01	3.4728E-01
10.00	6.5949E+04	-6.7144E+04	-2.9542E+05	9.2451E+04	-2.4263E-01	5.3000E-01	-6.2171E-01	4.7395E-01
9.75	-5.5026E+04	-1.2227E+04	2.3742E+05	3.3472E+04	-2.1122E-01	1.9988E-01	-5.5706E-01	-1.5128E-01
9.50	-1.1349E+05	3.4203E+04	2.9972E+05	-1.4262E+02	-2.8535E-01	-9.0025E-02	-4.3791E-01	-2.3955E-01
9.25	-1.4336E+05	6.9116E+04	4.0197E+05	-2.7771E+03	4.0721E-01	1.6271E-02	-4.0499E-01	-2.6189E-02
9.00	-1.5916E+05	6.7628E+04	4.0545E+05	-1.7404E+04	4.7888E-01	-1.2064E-01	-2.4824E-01	-9.9814E-02
8.75	-1.4929E+05	8.1233E+04	3.5037E+05	-3.9535E+04	5.4818E-01	-3.5712E-01	-9.5127E-02	1.4733E-01

Basic data, x = 25.0 mm.

y	Vt	theta	VO	V1	var0	var1	s01
30.00	626.37	110.03	588.48	-214.53	116.729	112.293	-61.55
21.00	582.87	102.00	570.12	-121.23	155.986	1425.719	-150.71
20.00	570.86	98.36	564.79	-83.02	122.455	1089.978	-183.06
19.50	569.84	97.80	564.57	-77.36	123.037	1127.853	-184.15
19.00	568.04	97.67	562.95	-75.83	116.420	1092.864	-122.31
18.50	568.34	97.34	563.68	-72.64	127.254	1287.951	-194.44
18.00	563.44	96.28	560.06	-61.64	313.597	1293.862	-246.48
17.50	559.09	95.54	556.48	-53.96	636.543	1329.822	-479.81
17.00	553.18	95.34	550.78	-51.44	1232.820	1755.841	-843.11
16.50	542.42	94.90	540.44	-46.34	2954.674	2145.133	-1632.37
16.00	533.09	94.43	531.49	-41.20	3425.741	1713.136	-1468.81
15.50	494.33	92.88	493.71	-24.85	6781.567	1857.436	-2372.84
15.00	466.01	92.80	465.46	-22.73	8955.245	1897.946	-2654.63
14.50	425.61	91.99	425.36	-14.75	11422.480	2308.966	-3540.95
14.00	395.42	91.44	395.30	-9.93	10095.002	2461.667	-3628.77
13.50	370.51	90.98	370.45	-6.35	10252.772	2483.062	-3573.02
13.00	349.15	90.94	349.10	-5.73	10481.388	2760.578	-3859.39
12.50	327.70	90.66	327.68	-3.79	9921.190	2886.565	-3720.75
12.00	287.47	91.72	287.34	-8.64	9915.507	2609.335	-3329.53
11.50	262.50	90.60	262.49	-2.74	8279.528	2605.469	-3027.36
11.25	256.57	90.98	256.53	-4.41	8133.522	2522.247	-2978.68
11.00	243.29	90.19	243.28	-0.81	7449.190	2347.077	-2612.77
10.75	242.05	90.94	242.02	-3.97	6912.335	2183.193	-2368.04
10.50	242.77	90.98	242.74	-4.16	6219.900	2228.734	-2177.42

Higher order moment data, x = 25.0 mm.

y	s001	s110	s000	s111	sk0	sk1	ku0	ku1
30.00	1.0848E+03	-9.0001E+02	-1.4515E+03	8.7687E+02	-1.1509E+00	7.3690E-01	3.8514E+00	2.6182E+00
21.00	-1.0916E+03	-5.6806E+01	-2.8102E+03	-3.2378E+04	-1.4425E+00	-6.0145E-01	8.9587E+00	-5.9189E-02
20.00	-1.0516E+03	4.7604E+03	1.9878E+02	-3.9193E+04	-1.4669E-01	-1.0891E+00	3.2863E-01	9.2733E-01
19.50	-2.5350E+02	4.1905E+03	-1.1169E+03	-5.2028E+04	-8.1840E-01	-1.3736E+00	5.0109E+00	2.6084E+00
19.00	-4.9211E+02	7.3092E+01	-1.0166E+03	-5.2028E+04	-8.0827E-01	-1.4400E+00	4.6724E+00	3.2664E+00
18.50	-8.4431E+02	7.1117E+03	-1.7772E+02	-6.7371E+04	-1.2380E-01	-1.4407E+00	1.9550E+00	2.9015E+00
18.00	1.1772E+04	-5.3387E+03	-2.2419E+04	-5.9449E+04	-4.0370E+00	-1.2774E+00	3.3174E+01	4.1372E+00
17.50	4.9437E+04	-3.1479E+04	-7.3126E+04	-3.6655E+04	-4.5534E+00	-7.5587E-01	2.0177E+01	4.8896E+00
17.00	8.3165E+04	-4.2205E+04	-1.5214E+05	-3.6103E+04	-3.5147E+00	-4.9070E-01	1.4415E+01	2.9889E+00
16.50	2.3224E+05	-1.0222E+05	-5.2541E+05	-2.0284E+05	-3.2963E+00	-2.0416E-01	1.2181E+01	2.8638E+00
16.00	1.8708E+05	-6.9380E+04	-8.9539E+05	-2.0070E+04	-2.6699E+00	-2.8305E-01	7.9896E+00	2.9946E+00
15.50	2.4058E+05	-1.0197E+05	-5.2534E+05	-4.8020E+04	-1.6033E+00	5.5966E-01	2.4774E+00	1.9726E+00
15.00	2.5415E+05	-1.0237E+05	-1.0237E+06	3.7822E+04	-1.2080E+00	4.5743E-01	1.1223E+00	1.4433E+00
14.50	1.9004E+05	-7.2911E+04	-8.3207E+05	3.4352E+04	-6.8158E-01	3.0998E-01	-4.1959E-01	1.0799E+00
14.00	1.1536E+05	-4.3950E+04	-3.6914E+05	5.0609E+04	-3.6394E-01	4.1437E-01	-5.8653E-01	5.1758E-01
13.50	6.8860E+04	-6.6587E+04	-2.5143E+05	6.0382E+04	-2.4219E-01	4.8800E-01	-6.8628E-01	3.1534E-01
13.00	4.1469E+04	-4.3950E+04	-1.6556E+05	4.9190E+04	-1.5429E-01	3.3914E-01	-7.3506E-01	3.3524E-01
12.50	-3.8953E+04	-1.4391E+04	8.8186E+03	4.3798E+04	8.9239E-03	2.8241E-01	-6.4540E-01	5.4739E-03
12.00	-1.0579E+05	1.3224E+04	3.3166E+05	1.6363E+04	3.3590E-01	1.2277E-01	-5.6003E-01	6.3626E-02
11.50	-1.4079E+05	5.1260E+04	3.9152E+05	-1.3699E+04	5.1970E-01	-1.0300E-01	-2.3951E-01	6.0997E-02
11.25	-1.3313E+05	5.0441E+04	4.0582E+05	-1.6867E+04	5.5325E-01	-1.3315E-01	-1.3291E-01	-1.8215E-01
11.00	-1.2533E+05	5.1658E+04	3.6842E+05	-1.7805E+04	5.7303E-01	-1.5658E-01	5.1980E-03	-6.7729E-02
10.75	-1.3458E+05	6.7742E+04	3.6316E+05	-2.9108E+04	6.3191E-01	-2.8535E-01	1.7596E-01	2.5672E-01
10.50	-1.3023E+05	7.0130E+04	3.2338E+05	-3.7301E+04	6.5923E-01	-3.5451E-01	4.3812E-01	2.7816E-01

Basic data, x = 30.0 mm.

y	Vt	theta	VO	V1	var0	var1	s01
35.00	624.48	109.98	586.90	-213.34	129.288	125.705	-69.78
24.00	579.99	101.54	568.27	-116.02	143.351	1552.255	-172.52
23.00	571.95	99.36	564.34	-92.99	130.643	1521.529	-189.30
22.00	567.87	98.11	562.18	-80.16	129.090	1504.931	-166.66
21.00	562.10	96.01	559.01	-58.85	125.275	928.417	-164.27
20.50	559.95	95.53	557.35	-53.92	160.173	985.826	-144.43
20.00	558.72	95.60	556.06	-54.49	244.115	1306.696	-233.82
19.50	557.56	96.46	554.02	-62.69	784.258	2502.344	-689.92
19.00	551.23	96.49	547.70	-62.30	1904.883	3403.986	-1418.14
18.50	542.59	94.88	540.63	-46.14	1999.413	2489.680	-1275.58
18.00	496.65	92.19	496.28	-18.95	7133.023	2451.691	-2877.18
17.50	478.02	92.21	477.67	-18.41	8306.979	2661.572	-3114.20
17.00	454.17	91.57	454.00	-12.43	10484.967	2334.930	-3267.19
16.50	422.31	91.09	422.23	-8.01	10159.162	2457.900	-3625.82
16.00	375.68	88.85	375.61	7.53	9991.463	2650.711	-3580.15
15.50	365.44	89.56	365.43	2.78	10246.301	2732.756	-3765.09
15.00	358.30	90.07	358.30	-0.43	9307.491	2465.418	-3468.33
14.50	335.83	89.96	335.83	0.25	9540.713	2578.665	-3416.04
14.00	319.01	89.96	319.01	0.22	8714.416	2658.002	-3243.98
13.50	302.71	89.60	302.70	2.10	7788.780	2472.115	-2764.58
13.25	280.01	90.97	279.97	-4.76	7707.948	2328.527	-2477.17
13.00	264.63	90.31	264.62	-1.44	6833.139	2307.003	-2270.97
12.75	255.88	89.94	255.88	0.26	6430.853	2051.967	-2044.61
12.50	254.14	90.49	254.14	-2.19	6046.320	1999.390	-1909.44

Higher order moment data, x = 30.0 mm.

y	s001	s110	s000	s111	sk0	sk1	ku0	ku1
35.00	9.7269E+02	-7.1704E+02	-1.3465E+03	6.7097E+02	-9.1592E-01	4.7607E-01	2.3876E+00	7.1204E-01
24.00	-8.7636E+02	4.9649E+02	-1.7122E+03	-4.5565E+04	-9.9758E-01	-7.4505E-01	5.4955E+00	5.4545E-02
23.00	-1.6812E+03	6.2221E+03	-5.8093E+02	-7.3279E+04	-3.8904E-01	-1.2347E+00	3.6625E+00	1.0673E+00
22.00	-1.8449E+03	9.0039E+03	-9.7353E+02	-9.8014E+04	-6.6376E-01	-1.6789E+00	4.2567E+00	2.6820E+00
21.00	-5.9858E+02	8.0793E+03	-4.1297E+01	-5.6681E+04	-2.9452E-02	-2.0037E+00	2.2962E+00	5.8606E+00
20.50	-5.9273E+02	5.1969E+03	-2.3921E+03	-5.7164E+04	-1.1801E+00	-1.8468E+00	7.3663E+00	6.1293E+00
20.00	1.5680E+03	6.9594E+03	-6.5136E+03	-7.7780E+04	-1.7078E+00	-1.6467E+00	8.5012E+00	4.3415E+00
19.50	6.2473E+04	-3.2793E+04	-8.8890E+04	-1.1277E+05	-4.0473E+00	-9.0092E-01	2.2662E+01	1.3627E+00
19.00	1.8856E+05	-9.2939E+04	-3.3020E+05	-8.0955E+04	-3.9717E+00	-4.0763E-01	1.9233E+01	7.7420E-01
18.50	1.1253E+05	-4.4955E+04	-2.4063E+05	-9.2263E+04	-2.6915E+00	-7.4270E-01	7.9906E+00	2.2924E+00
18.00	3.3004E+05	-1.1783E+05	-9.8986E+05	5.7806E+03	-1.6431E+00	4.7618E-02	2.5115E+00	1.9143E+00
17.50	2.5615E+05	-7.0844E+04	-9.3344E+05	-3.5639E+04	-1.2329E+00	-2.5955E-01	9.3274E-01	1.5553E+00
17.00	2.3120E+05	-7.7147E+04	-9.9893E+05	1.2785E+04	-9.3044E-01	1.1332E-01	-7.1357E-02	1.3159E+00
16.50	1.1995E+05	-4.0623E+04	-4.7857E+05	1.0497E+04	-4.6736E-01	8.6147E-02	-7.1542E-01	7.0695E-01
16.00	3.0521E+04	-5.2054E+04	-1.6952E+05	5.0908E+04	-1.6974E-01	3.7903E-01	-7.4556E-01	1.3337E-01
15.50	2.2123E+04	-4.6001E+04	-9.2314E+04	4.7953E+04	-8.9005E-02	3.3567E-01	-7.9542E-01	6.0550E-02
15.00	-5.1464E+03	-1.7339E+04	-4.2407E+04	2.9160E+04	-4.7227E-02	2.3821E-01	-7.9392E-01	7.3480E-02
14.50	-6.8719E+04	1.0784E+04	1.0102E+05	1.6315E+04	1.0840E-01	1.2452E-01	-8.0221E-01	-2.2148E-01
14.00	-7.8654E+04	1.6350E+04	1.5907E+05	1.4025E+04	1.9554E-01	1.0234E-01	-6.8529E-01	-2.0309E-01
13.50	-9.0231E+04	3.2244E+04	1.9470E+05	-3.9661E+03	2.8325E-01	-3.2267E-02	-4.9703E-01	-1.5099E-01
13.25	-1.3454E+05	4.9733E+04	3.6098E+05	-1.4454E+04	5.3343E-01	-1.2864E-01	-2.3624E-01	-1.3302E-02
13.00	-1.4196E+05	6.0094E+04	3.9046E+05	-1.9564E+04	6.9127E-01	-1.7656E-01	1.3297E-01	1.3265E-01

12.75	-1.2296E+05	6.3514E+04	3.2242E+05	-3.0446E+04	6.2519E-01	-3.2755E-01	2.2083E-01	1.1914E-01
12.50	-9.8929E+04	5.7439E+04	2.9082E+05	-3.3490E+04	6.1858E-01	-3.7460E-01	1.4764E-01	1.8168E-01

Basic data, x = 35.0 mm.

y	Vt	theta	VO	V1	var0	var1	s01
35.00	624.59	110.05	586.74	-214.13	104.275	122.313	-46.92
27.00	570.76	99.95	562.18	-98.63	130.849	1713.864	-230.35
26.00	566.15	98.46	560.00	-83.25	142.465	1475.569	-195.48
25.00	564.98	97.84	559.70	-77.05	152.145	1774.390	-207.48
24.00	559.96	96.39	556.48	-62.34	126.394	1716.906	-188.26
23.00	557.41	95.49	554.85	-53.36	152.616	1281.892	-179.61
22.50	554.63	94.79	552.70	-46.29	235.107	1315.710	-211.62
22.00	550.83	94.96	548.76	-47.66	969.095	2147.524	-686.17
21.50	549.09	94.52	547.39	-43.25	1160.417	2213.936	-800.86
21.00	543.60	94.13	542.19	-39.11	1849.980	2328.756	-1151.68
20.50	533.19	94.00	531.89	-37.18	3342.092	3115.914	-2057.17
20.00	514.75	92.37	514.31	-21.25	4210.062	1899.129	-1825.16
19.50	498.74	92.06	498.41	-17.96	5584.182	1828.731	-1997.69
19.00	467.22	91.61	467.03	-13.16	7993.199	2348.696	-2800.77
18.50	442.71	90.48	442.69	-3.73	9992.555	2261.464	-3178.73
18.00	413.25	90.64	413.23	-4.61	8873.327	2409.175	-3257.43
17.50	387.03	89.66	387.02	2.30	9124.133	2426.867	-3296.79
17.00	372.22	89.43	372.20	3.73	8740.143	2489.362	-3251.64
16.50	362.23	89.98	362.23	0.14	8462.497	2501.807	-3142.15
16.00	337.21	89.13	337.17	5.12	7849.018	2513.630	-2861.06
15.50	325.46	89.15	325.42	4.85	6912.124	2459.199	-2657.15
15.25	318.00	89.08	317.96	5.12	6655.475	2286.293	-2330.65
15.00	310.95	88.99	310.91	5.49	6396.124	2134.903	-2232.57
14.75	305.06	89.11	305.02	4.72	5943.905	2119.469	-2087.01
14.50	297.82	88.72	297.75	6.65	5708.782	2081.372	-1905.93
14.25	294.82	88.78	294.75	6.30	5694.945	1858.643	-1690.62

Higher order moment data, x = 35.0 mm.

y	s001	s110	s000	s111	sk0	sk1	ku0	ku1
35.00	5.5544E+02	-4.7868E+02	-8.4888E+02	3.8190E+02	-7.9722E-01	2.8232E-01	1.7099E+00	2.6051E-01
27.00	-1.6641E+03	5.9972E+03	1.4294E+02	-6.537E+04	9.5496E-02	-9.3777E-01	2.2270E+00	3.9760E-01
26.00	-2.6085E+03	8.0258E+03	-5.3078E+02	-7.6056E+04	-3.1214E-01	-1.3418E+00	5.0774E+00	1.3055E+00
25.00	-2.3016E+03	7.9948E+03	9.8321E+02	-1.0809E+05	5.2392E-01	-1.4461E+00	5.0227E+00	1.2430E+00
24.00	-1.6154E+03	1.0259E+04	2.3172E+02	-1.2188E+05	1.6307E-01	-1.7133E+00	2.9607E+00	2.5792E+00
23.00	-8.4129E+02	7.7841E+03	-1.2443E+03	-7.7126E+04	-6.5998E-01	-1.6804E+00	4.8262E+00	3.7140E+00
22.50	7.7657E+03	-2.2230E+03	-8.2381E+03	-8.5458E+04	-2.2852E+00	-1.7907E+00	2.0185E+01	6.4185E+00
22.00	7.1235E+04	-2.4640E+04	-1.4792E+05	-9.7724E+04	-4.9033E+00	-9.8197E-01	3.2799E+01	2.5390E+00
21.50	8.8429E+04	-3.5974E+04	-1.6941E+05	-1.0515E+05	-4.2856E+00	-1.0094E+00	2.4575E+01	2.9380E+00
21.00	1.3737E+05	-4.6353E+04	-2.8598E+05	-9.6787E+04	-3.5940E+00	-8.6125E-01	1.6091E+01	2.4603E+00
20.50	2.6478E+05	-1.1029E+05	-5.3800E+05	-7.1411E+04	-2.7845E+00	-4.1057E-01	8.0967E+00	1.5048E+00
20.00	1.9883E+05	-7.8660E+04	-5.7311E+05	-2.9476E+03	-2.0980E+00	-3.5616E-02	4.8795E+00	3.0404E+00
19.50	1.9652E+05	-7.5946E+04	-6.6516E+05	5.9358E+03	-1.5940E+00	7.5902E-02	2.2496E+00	2.5017E+00
19.00	1.8106E+05	-4.6635E+04	-6.8592E+05	-3.6030E+04	-9.5983E-01	-3.1654E-01	1.8418E-01	1.7440E+00
18.50	1.2622E+05	-3.1055E+04	-6.3440E+05	-3.6224E+02	-6.3511E-01	-3.3683E-03	-6.5227E-01	1.1997E+00
18.00	5.0434E+04	-2.8590E+04	-2.4590E+05	7.8855E+03	-2.9419E-01	6.6685E-02	-7.9999E-01	7.5791E-01
17.50	4.3623E+04	-4.9842E+04	-1.3176E+05	4.2969E+04	-1.5118E-01	3.5941E-01	-8.2296E-01	1.2963E-02
17.00	-7.8561E+03	-2.4856E+04	-2.4932E+04	4.3433E+04	-3.0512E-02	3.4970E-01	-8.2246E-01	1.2451E-01
16.50	-1.4774E+04	-1.4780E+04	-2.5431E+04	2.8099E+04	-3.2667E-02	2.2455E-01	-6.4084E-01	5.2729E-02
16.00	-1.0476E+05	3.5176E+04	1.9640E+05	-3.0202E+03	2.8243E-01	-2.3965E-02	-6.3772E-01	4.8769E-02
15.50	-9.9212E+04	3.5533E+04	2.0905E+05	-3.8544E+02	3.6377E-01	-3.1606E-03	-4.0936E-01	-1.6722E-01
15.25	-1.1035E+05	4.6696E+04	1.9052E+05	-4.2296E+03	3.5089E-01	-3.8690E-02	-3.8998E-01	-1.5572E-02
15.00	-1.1864E+05	6.3450E+04	2.0898E+05	-1.7303E+04	4.0854E-01	-1.7541E-01	-2.6187E-01	1.8947E-01
14.75	-1.066E+05	6.0815E+04	1.9599E+05	-1.7205E+04	4.2770E-01	-1.9682E-01	-1.3925E-01	3.1190E-01
14.50	-1.0545E+05	5.5177E+04	1.8243E+05	-2.0418E+04	4.2294E-01	-2.1503E-01	6.4236E-02	4.3302E-01
14.25	-8.7247E+04	5.5625E+04	1.5297E+05	-2.6684E+04	3.5594E-01	-3.3301E-01	-1.1583E-01	4.2253E-01

Basic data, x = 40.0 mm.

y	Vt	theta	VO	V1	var0	var1	s01
40.00	619.79	109.82	583.10	-210.10	160.320	182.309	-104.23
29.00	560.67	97.67	555.65	-74.82	139.489	1806.798	-284.33
28.00	561.47	98.31	555.58	-81.14	254.355	2700.398	-398.69
27.00	554.82	95.19	552.55	-50.23	118.039	929.978	-138.67
26.00	556.79	95.56	554.17	-53.90	162.366	1626.057	-278.28
25.00	553.98	94.66	552.15	-45.03	181.133	1539.031	-239.81
24.50	551.96	94.29	550.42	-41.29	229.265	1288.827	-237.17
24.00	549.03	94.80	547.10	-45.99	878.172	2446.083	-818.25
23.50	544.12	94.65	542.33	-44.14	1880.232	2976.443	-1356.45
23.00	533.08	93.62	532.02	-33.67	3495.716	2723.742	-1743.19
22.50	526.06	93.80	524.91	-34.82	4116.333	3465.048	-2464.43
22.00	512.90	91.84	512.64	-16.50	4196.119	1606.409	-1712.48
21.50	505.23	91.70	505.01	-15.01	4246.289	1381.263	-1491.23
21.00	478.59	91.32	478.46	-11.04	6529.839	1759.626	-2159.98
20.50	464.82	91.54	464.65	-12.45	6774.542	1711.446	-2028.38
20.00	412.31	89.80	412.31	1.46	9337.557	2167.280	-3086.90
19.50	376.83	88.52	376.70	9.74	9282.037	2341.569	-2976.25
19.00	357.26	87.79	356.99	13.80	8143.571	2412.743	-2889.91
18.50	342.47	87.76	342.21	13.41	7156.485	2280.957	-2542.66
18.00	328.48	87.44	328.15	14.68	6532.400	2296.314	-2392.16
17.50	315.11	87.68	314.85	12.73	5720.797	2012.182	-1964.18
17.00	308.10	87.65	307.90	11.00	5438.109	1939.306	-1792.17
16.75	302.80	87.99	302.61	10.61	5031.869	1894.254	-1620.49
16.50	299.45	87.82	299.23	11.38	4677.565	1724.155	-1419.77
16.25	298.18	88.30	298.05	8.84	4569.933	1603.462	-1262.32
16.00	296.28	88.96	296.23	5.36	4656.351	1560.909	-1308.31

Higher order moment data, x = 40.0 mm.

y	s001	s110	s000	s111	sk0	sk1	ku0	ku1
40.00	2.7865E+03	-2.6980E+03	-3.0090E+03	2.7714E+03	-1.4823E+00	1.1259E+00	5.4265E+00	4.2176E+00
29.00	-3.0522E+03	1.3991E+04	5.0491E+02	-1.0102E+05	3.0648E-01	-1.3153E+00	1.7816E-01	1.0299E+00
28.00	-6.6475E+03	1.5742E+04	5.6858E+02	-1.2707E+05	1.4016E-01	-9.0554E-01	4.3650E+00	-4.3938E-01
27.00	-2.2949E+03	1.0293E+04	4.2514E+02	-6.2722E+04	3.3151E-01	-2.2116E+00	8.8029E-01	6.0398E+00
26.00	-2.7956E+03	1.7665E+04	3.4196E+02	-1.2058E+05	1.6528E-01	-1.8389E+00	2.3652E+00	3.0068E+00
25.00	1.5204E+03	1.0191E+04	-2.7967E+03	-9.4839E+04	-1.1472E+00	-1.5708E+00	9.7585E+00	4.1478E+00
24.50	2.9443E+03	7.4575E+03	-5.8891E+03	-8.1598E+04	-1.6965E+00	-1.7635E+00	9.5235E+00	5.2612E+00
24.00	7.8211E+04	-3.9038E+04	-1.1114E+05	-9.8057E+04	-4.2707E+00	-8.1053E-01	2.7039E+01	2.0043E+00
23.50	1.5293E+05	-6.8416E+04	-2.7849E+05	-9.6991E+04	-3.4158E+00	-5.9729E-01	1.3894E+01	1.5195E+00
23.00	2.3682E+05	-7.7289E+04	-6.4013E+05	-8.8036E+04	-3.0972E+00	-6.1932E-01	1.0321E+01	2.0140E+00
22.50	2.9622E+05	-1.0824E+05	-6.4967E+05	-7.8669E+04	-2.4600E+00	-3.8569E-01	6.3796E+00	1.1810E+00
22.00	1.7812E+05	-6.6396E+04	-5.4704E+05	7.5968E+03	-2.0125E+00	1.1799E-01	4.2381E+00	2.5022E+00
21.50	1.5097E+05	-6.0458E+04	-4.8099E+05	1.5603E+04	-1.7383E+00	3.0394E-01	2.8560E+00	2.9779E+00
21.00	1.6367E+05	-6.2344E+04	-5.9533E+05	1.6192E+04	-1.1282E+00	2.1937E-01	6.2070E-01	1.8563E+00

20.50	1.2495E+05	-6.7392E+04	-4.8755E+05	2.7115E+04	-8.7437E-01	3.8297E-01	-2.7690E-03	1.5246E+00
20.00	3.4870E+04	-2.6981E+04	-2.2637E+05	2.6174E+04	-2.5088E-01	2.5942E-01	-9.6050E-01	4.1185E-01
19.50	-3.9356E+04	2.7878E+03	2.0880E+04	1.2684E+04	2.3349E-02	1.1194E-01	-8.8203E-01	1.7675E-01
19.00	-7.3300E+04	8.3215E+03	1.6108E+05	2.0135E+04	2.1919E-01	1.6990E-01	-7.8683E-01	-1.9345E-01
18.50	-8.1488E+04	1.9015E+04	1.8971E+05	1.0973E+04	3.1335E-01	1.0073E-01	-5.2962E-01	-6.8673E-02
18.00	-9.5609E+04	3.7144E+04	2.2273E+05	-7.2389E+03	4.1993E-01	-6.5785E-02	-2.7256E-01	1.2985E-01
17.50	-1.0222E+05	4.1870E+04	2.1894E+05	-1.4725E+04	5.0598E-01	-1.6314E-01	-1.0208E-01	9.4443E-02
17.00	-8.5101E+04	4.1269E+04	1.9546E+05	-1.5034E+04	4.8740E-01	-1.7604E-01	-9.4057E-02	-2.3280E-02
16.75	-9.2866E+04	5.1063E+04	1.9740E+05	-2.3349E+04	5.5306E-01	-2.8321E-01	1.8546E-01	5.7218E-01
16.50	-7.1653E+04	4.3243E+04	1.6464E+05	-2.0968E+04	5.1464E-01	-2.9289E-01	5.6312E-02	4.7952E-01
16.25	-6.4395E+04	4.0316E+04	1.5437E+05	-1.8122E+04	4.9967E-01	-2.8224E-01	1.1689E-01	5.5596E-01
16.00	-6.7313E+04	4.5256E+04	1.5913E+05	-1.9708E+04	5.0082E-01	-3.1958E-01	7.7204E-02	9.0249E-01

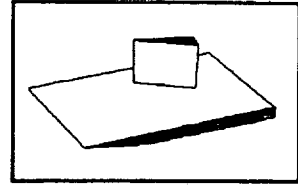
Ref.: 3-6

Author: Kussoy and Horstman

Geometry: 3-D Fin

Mach number: 8.2

Data: p_{wall} , q_{wall} , c_f , flowfield pitot surveys

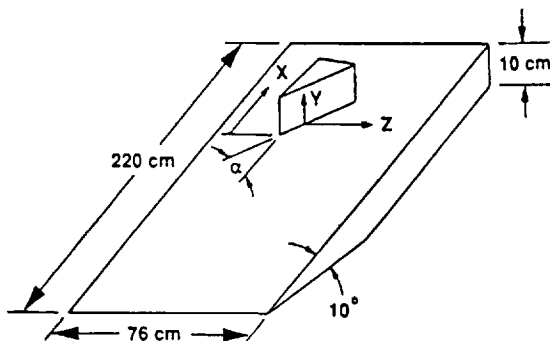


A detailed investigation of hypersonic sharp-fin interactions on a flat plate was carried out by M. I. Kussoy, K. C. Horstman, and C. C. Horstman of NASA-Ames Research Center (Refs. 3-6). Since a detailed data report (Ref. 3) has already been prepared on this experiment, the present treatment will be brief. The data included in the file KUSSOY3.DAT are exactly those given in Ref. 3 for sharp-fin interactions, plus some additional data (notably skin friction) supplied by the experimenters after Ref. 3 was published.

The data were obtained in the NASA-Ames 3.5-foot hypersonic wind tunnel, using a flat-plate model with a length of 220 cm. Natural transition occurred on this plate for all test conditions. While no turbulence data are available to establish the condition of the incoming boundary layer ahead of the interaction, a mean profile included in this dataset shows typical law-of-the-wall behavior, albeit with a lower wake-strength parameter than usual.

Fins of 5, 7.5, 10, 12.5, and 15 degree angle-of-attack were mounted on the flat plate. The fin leading-edge position was 176 cm aft of the plate leading-edge. The x,y,z coordinate frame used to describe the measurements is indicated in the sketch below. Given this information and the table of freestream properties in the data file, most of the measured distributions are self-explanatory.

One feature of this experiment is redundant data, considered a desirable aspect of benchmark experiments in Ref. 1. Similar heat transfer data are obtained by two different techniques. Redundant pressure and heat transfer data are included in the data file to establish the repeatability of the experiment.



Users of these data should note that this fin interaction involves a relatively-thicker incoming boundary layer than the supersonic fin interactions included elsewhere in this database. The reason for this is the long flat-plate run required to naturally establish a turbulent boundary-layer at Mach 8. Thus the present fin is only 8 incoming-boundary-layer-thicknesses long, compared to relative fin lengths 5 times larger in the Penn State experiments at Mach 3 and 4. The net effect is that the present interaction lies entirely within its turbulent boundary-layer, while the supersonic interactions just referred to extend far outside their respective boundary-layers. The supersonic and Mach 8 cases are thus not directly comparable.

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*****
KUSOYS3.DAT
*****
DATA FILES FROM NASA TM 103838 AND SUPPLEMENTARY DATA SUPPLIED
BY THE EXPERIMENTERS ON A SINGLE-FIN, SHEPT SHOCK WAVE/
TURBULENT BOUNDARY-LAYER INTERACTION AT MACH 8.2
*****
FREESTREAM CONDITIONS
*****
H INF      = 8.18
T INF      = 81 K
P INF      = 430 N/M^2
TM         = 300 K
U INF      = 1446 M/S
DELTA* O   = 3.7 CM
THETA O    = 1.59 CM
THETA O    = 0.09% CM
TAUM INF   = 19.6 N/M^2
Q INF      = 9000 W/M^2
RE/M       = 4.9EE+06
CF INF     = 0.98EE-03
CH INF     = 0.53EE-03
*****
S = DISTANCE ALONG FIN SURFACE MEASURED FROM LEADING EDGE
*****
UPSTREAM BOUNDARY LAYER
*****

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Y(cm)	H	P / P INF	RHO / RHO INF	T / T INF	U / U INF	RHO / RHO INF	TT / TT INF
0.000	0.000	1.000	0.270	3.699	0.000	0.000	0.270
0.070	1.777	1.000	0.213	4.705	0.481	0.102	0.555
0.140	2.069	1.000	0.195	5.138	0.584	0.114	0.682
0.200	2.647	1.000	0.237	4.217	0.678	0.161	0.721
0.280	3.083	1.000	0.266	3.756	0.746	0.199	0.773
0.360	3.409	1.000	0.295	3.390	0.784	0.231	0.798
0.430	3.558	1.000	0.301	3.323	0.810	0.244	0.828
0.500	3.747	1.000	0.333	3.002	0.811	0.270	0.808
0.710	4.068	1.000	0.345	2.897	0.864	0.298	0.877
0.920	4.422	1.000	0.386	2.593	0.889	0.343	0.894
1.120	4.750	1.000	0.419	2.368	0.916	0.394	0.922
1.320	5.106	1.000	0.453	2.205	0.947	0.429	0.956
1.520	5.461	1.000	0.504	1.982	0.960	0.484	0.963
1.720	5.774	1.000	0.560	1.785	0.963	0.539	0.956
1.920	6.101	1.000	0.600	1.668	0.964	0.590	0.981
2.130	6.411	1.000	0.671	1.490	0.977	0.656	0.959
2.320	6.689	1.000	0.705	1.419	0.995	0.701	0.983
2.510	7.009	1.000	0.768	1.302	0.998	0.767	0.981
2.710	7.246	1.000	0.820	1.220	0.999	0.819	0.977
3.040	7.617	1.000	0.865	1.156	1.023	0.884	1.012
3.380	7.978	1.000	0.944	1.060	1.025	0.968	1.011
3.720	8.180	1.000	1.000	1.000	1.021	1.021	1.000
4.070	8.180	1.000	0.995	1.005	1.024	1.019	1.004

SURFACE STREAMLINE ANGLES ON FLAT PLATE WITH FIN

10 DEGREE FIN ANGLE	15 DEGREE FIN ANGLE
Z(cm)	Z(cm)
(deg)	(deg)
0.20	10.0
0.40	15.5
0.84	21.3
1.28	29.4
1.83	34.3
	0.22
	0.68
	1.20
	1.75
	2.12
	35.0
	35.7
	44.7
	47.2
	50.3

2.22	34.8	2.44	50.5
2.63	35.2	2.85	45.2
2.96	30.0	3.30	39.0
3.28	25.0	3.70	27.2
4.30	24.0	4.70	32.8
5.50	23.8	5.40	33.0
6.78	23.2	6.34	32.0
7.59	17.2	7.10	34.2
8.30	10.8	8.75	32.2
8.91	0.0	9.42	27.5
		10.15	7.5
		10.55	0.0

SURFACE PRESSURES ON FIN (P/P INF)

Y(cm)	Fin Angle = 5 deg	7.5 deg	10 deg	12.5 deg	15 deg
0.45	1.855	2.565	3.532	4.516	5.871
0.95	1.823	2.484	3.323	4.210	5.274
1.45	1.790	2.452	3.242	4.065	5.000
2.45	1.984	2.839	3.887	5.016	6.484
3.43	2.145	3.290	4.823	6.242	8.306
4.43	2.355	3.613	5.371	7.000	9.355
5.42	2.419	3.774	5.565	7.274	9.677
6.42	2.435	3.855	5.677	7.435	9.839
7.42	2.516	3.903	5.742	7.532	9.948
8.41	2.597	4.032	5.839	7.694	10.161
9.40	2.565	4.016	5.935	7.774	10.306
10.40	2.613	4.145	6.065	7.887	10.468

SURFACE PRESSURES ON FLAT PLATE WITH FIN X = 18.19 cm

Fin Angle =	5 deg	7.5 deg	10 deg
Z(cm)	P/P INF	P/P INF	P/P INF
15.82	1.000	15.29	1.089
14.82	0.979	14.29	1.050
13.82	0.968	13.29	1.061
12.83	0.990	12.30	1.050
11.83	0.940	11.30	1.044
10.80	1.027	10.27	1.103
9.82	0.994	9.29	1.065
8.82	0.990	8.29	1.123
7.82	1.008	7.29	1.197
6.82	1.115	6.29	1.358
5.82	1.169	5.29	1.506
4.82	1.324	4.29	1.661
3.83	1.410	3.30	1.774
2.83	1.497	2.30	1.935
1.83	1.608	1.30	2.371
0.83	2.000	0.30	2.887

Fin Angle = 12.5 deg

15 deg	Z(cm)	P/P INF
14.27	1.229	13.72
13.27	1.213	12.72
12.27	1.226	11.72
11.27	1.037	10.73
10.27	1.223	9.73
9.25	1.252	8.70
8.27	1.327	7.72
7.27	1.500	6.72
6.27	1.726	5.72
	1.968	2.165
		2.339

5.27	2.194	4.72	2.452	15.75	0.86	15.27	0.95	14.76	0.99
4.27	2.371	3.72	2.452	14.77	0.96	14.29	0.99	13.78	1.02
3.27	2.587	2.72	2.613	13.76	0.94	13.28	1.06	12.77	1.05
2.28	2.758	1.73	4.113	12.78	0.95	12.30	1.03	11.79	1.05
1.28	3.984	0.73	6.387	11.82	0.92	11.34	1.01	10.83	1.07
0.28	5.065			10.83	0.96	10.35	1.06	9.84	1.08
				9.82	0.98	9.34	1.10	8.83	1.07
				8.82	0.98	8.34	1.03	7.83	1.04
				8.80	0.96	8.32	1.03	7.81	1.20
				5.80	1.05	5.32	1.18	4.81	1.58
				4.80	1.11	4.32	1.30	3.81	1.58
				2.54	1.29	2.06	1.87	1.55	2.61
				1.79	1.47	1.31	2.31	0.80	3.23

Additional Surface Pressures on Flat Plate with Fin
x = 18.19cm

10 deg fin angle			15 deg fin angle		
z(cm)	P/PINF	z(cm)	z(cm)	P/PINF	z(cm)
11.50	0.95	5.49	1.78	13.49	0.96
11.02	0.99	5.04	1.84	12.99	0.96
10.47	0.96	4.49	1.97	12.50	0.96
10.04	1.00	4.05	1.99	12.00	0.96
9.49	1.00	3.50	2.05	11.50	1.01
9.04	1.07	3.05	1.98	11.00	1.04
8.49	1.13	2.50	2.21	10.47	1.20
8.04	1.21	2.05	2.48	9.97	1.26
7.49	1.32	1.50	3.00	9.49	1.46
7.04	1.41	1.05	3.42	8.99	1.54
6.49	1.55	0.50	4.02	8.49	1.80
6.04	1.63	0.05	3.81	7.99	1.87
				7.49	2.12
				6.99	2.18

HEAT TRANSFER ON FIN (Q/Q INF)

Thermocouples, S = 16.70 cm

Y(cm)	Fin Angle = 5 deg	7.5 deg	10 deg	12.5 deg	15 deg
0.47	1.28	2.06	3.04	3.82	5.01
0.95	1.17	1.80	2.86	3.47	4.24
1.45	1.30	2.11	2.99	3.69	4.55
2.45	1.40	2.37	3.27	4.20	5.50
3.45	1.44	2.32	3.02	3.83	5.30
4.45	1.43	1.98	2.54	2.90	4.02
5.45	1.30	1.85	2.31	2.60	3.35
6.43	1.25	1.68	2.07	2.35	3.05
7.43	1.19	1.65	2.03	2.41	3.09
8.43	1.20	1.69	2.06	2.38	3.08
9.40	1.21	1.67	2.04	2.39	2.93
10.38	1.16	1.67	2.02	2.42	2.94

Schmidt-Boelter Gauges, S = 19.62 cm

Y(cm)	Fin Angle = 5 deg	7.5 deg	10 deg	12.5 deg	15 deg
2.45	1.34	1.65	2.61	3.43	4.28
3.45	1.39	1.64	2.61	3.54	4.74
4.43	1.43	1.64	2.40	3.12	4.26
5.42	1.30	1.51	2.18	2.77	3.69
7.15	1.06	1.23	1.74	2.31	3.01
9.15	1.17	1.38	1.91	2.57	3.26

HEAT TRANSFER ON FLAT PLATE WITH FIN

x = 16.45 cm

Fin Angle =	5 deg	7.5 deg	10 deg	12.5 deg	15 deg
Z(cm)	Q/Q INF	Z(cm)	Q/Q INF	Z(cm)	Q/Q INF

Fin Angle = 12.5 deg

Z(cm)	Q/Q INF	Z(cm)	Q/Q INF
14.35	0.99	13.89	0.98
13.37	1.01	12.91	1.00
12.36	1.02	11.90	1.03
11.38	1.01	10.92	1.00
10.42	1.02	9.96	0.94
9.43	1.06	8.97	0.95
8.42	1.00	7.96	0.95
7.42	0.95	6.96	1.46
6.40	1.58	4.94	2.01
5.40	1.85	3.94	1.80
4.40	1.56	2.94	1.75
3.40	1.14	1.94	3.34
2.40	0.59	0.94	4.07

Additional Heat Transfer on Flat Plate with Fin
x = 16.45cm

10 deg fin angle		15 deg fin angle	
Z(cm)	Q/QINF	Z(cm)	Q/QINF
11.87	1.02	5.85	1.33
11.43	1.08	5.40	1.46
10.88	1.06	4.85	1.54
10.43	1.06	3.14	1.68
9.88	1.02	2.59	1.95
9.42	1.08	2.39	2.10
8.87	1.01	1.84	2.51
7.40	1.02	1.39	2.98
6.85	1.08	0.84	3.18
6.40	1.17	0.35	3.11

Skin-friction on Flat Plate with Fin, x = 15.5cm
(replaces data in AIAA paper 91-1761 which was in error)

Fin angle = 10 deg			15 deg		
z(cm)	Cf x 10	uncertainty	z(cm)	Cf x 10	uncertainty
9.00	1.00	+/- 0.18	9.00	1.00	+/- 0.18
7.12	0.84	+/- 0.10	2.73	2.40	+/- 0.33
6.52	0.87	+/- 0.21	1.35	4.59	+/- 0.88
4.22	1.56	+/- 0.33	1.05	4.50	+/- 0.81
3.21	1.08	+/- 0.20	0.60	5.86	+/- 1.33
2.16	1.87	+/- 0.48			
1.24	2.92	+/- 0.54			
0.83	2.49	+/- 0.56			

FLOW FIELD YAW ANGLES (DEGREES)									
10 DEGREE FIN ANGLE									
X = 17.23 cm									
Y(cm)	Z(cm) = 0.64	1.27	1.91	2.54	3.81	5.08	6.35	7.62	
0.00									
0.25	16	25	35	35	25	24			
0.50	14	17	20	22	21	15			
0.75	14	15	18	19	15	9			
1.00	13	14	16	15	10	4			
1.25	13	13	14	11	5	2			
1.50	13	12	12	7	2	1			
1.75	12	11	10	5	1	1			
2.00	12	11	9	3	0	0			
2.25	12	10	9	2	0	0			
2.50	11	10	9	2	0	0			
2.75	11	10	9	2	0	0			
3.00	11	10	10	2	0	0			
3.25	11	10	11	2	0	0			

FLOW FIELD PITOT PRESSURES (PT2/PT2 INF)
10 DEGREE FIN ANGLE
X = 17.23 cm

Y(cm)	Z(cm) = 0.64	1.27	1.91	2.54	3.81	5.08
0.25	0.51	0.51	0.40	0.33	0.16	0.15
0.50	0.70	0.60	0.39	0.31	0.20	0.20
0.75	0.85	0.63	0.38	0.30	0.24	0.28
1.00	0.95	0.69	0.42	0.33	0.32	0.35
1.25	1.04	0.75	0.48	0.39	0.41	0.44
1.50	1.16	0.84	0.57	0.49	0.51	0.55
1.75	1.26	0.96	0.70	0.61	0.64	0.62
2.00	1.44	1.11	0.86	0.76	0.78	0.71
2.25	1.61	1.27	1.07	0.95	0.88	0.75
2.50	1.79	1.48	1.31	1.15	0.89	0.80
2.75	1.97	1.71	1.67	1.33	0.91	0.85
3.00	2.15	1.96	1.97	1.40	0.93	0.89
3.25	2.47	2.21	2.32	1.32	0.94	0.95

FLOW FIELD YAW ANGLES (DEGREES)
15 DEGREE FIN ANGLE
X = 17.44 cm

Y(cm)	Z(cm) = 1.27	1.91	2.54	3.81	5.08	6.35	7.62
0.00							
0.25	44	46	50	37	33	32	34
0.50	25	23	32	32	25	25	14
0.75	22	21	30	29	27	17	7
1.00	21	20	26	23	19	9	3
1.25	19	18	20	17	11	5	2
1.50	18	16	12	10	5	2	1
1.75	17	13	7	4	2	1	1
2.00	16	10	2	1	1	0	0
2.25	15	8	1	0	0	0	0
2.50	14	3	1	0	0	0	0
2.75	14	1	0	0	0	0	0
3.00	14	1	0	0	0	0	0
3.25	15	1	0	0	0	0	0

FLOW FIELD PITOT PRESSURES (PT2/PT2 INF)
15 DEGREE FIN ANGLE
X = 17.44 cm

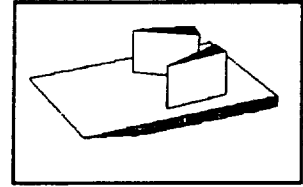
Ref.: 14

Author: Kussoy and Horstman

Geometry: Crossing Oblique Shock Waves

Mach number: 8.3

Data: p_{wall} , q_{wall} , flowfield pitot surveys

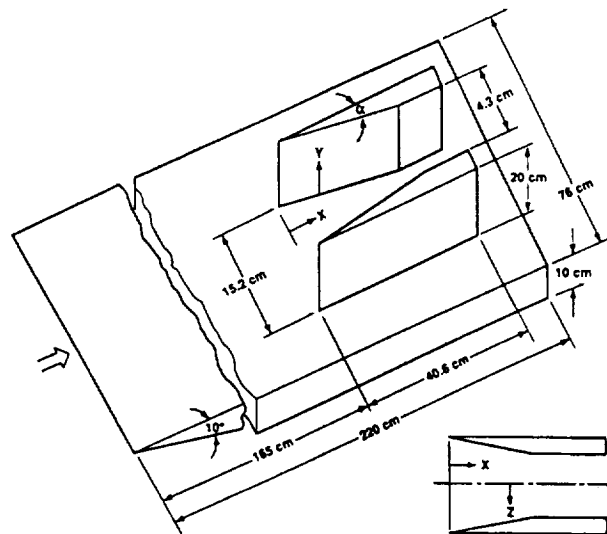


A detailed investigation of hypersonic crossing-shock interactions on a flat plate was carried out by M. I. Kussoy, K. C. Horstman, and C. C. Horstman of NASA-Ames Research Center (Refs. 14). Since a detailed data report (Ref. 14) has already been prepared on this experiment, the present treatment will be brief.

The data included in the file KUSSOY4.DAT are exactly those given in Ref. 14. These data were obtained in the NASA-Ames 3.5-foot hypersonic wind tunnel, using a flat-plate model with a length of 220 cm. Natural transition occurred on this plate for all test conditions. While no turbulence data are available to establish the condition of the incoming boundary layer ahead of the interaction, a mean profile included in this dataset shows typical law-of-the-wall behavior, albeit with a lower wake-strength parameter than usual.

Opposing pairs of fins, each having either 10 or 15 degree angles-of-attack were mounted on the flat plate as shown in the sketch below. The x,y,z coordinate frame used to describe the measurements is also indicated in the sketch. Given this information and the table of freestream properties in the data file, most of the measured distributions are self-explanatory.

As in the case of the Mach 8 single-fin interaction, Refs. 3-6, users of these data should note that this crossing-shock interaction involves a relatively-thicker incoming boundary layer than the supersonic crossing-shock interactions included elsewhere in this database. The supersonic and Mach 8 cases are thus not directly comparable.



 KUSOY4.DAT
 DATA FILES FROM NASA TM 103909
 ON A CROSSING-SHOCK-WAVE/
 TURBULENT BOUNDARY-LAYER INTERACTION AT MACH 8.3

 FREESTREAM CONDITIONS
 M INF = 8.28
 T INF = 80 K
 P INF = 430 N/M^2
 RHO INF = 0.0186 KG/M^3
 MU INF = 300 K
 U INF = 1483 M/S
 DELTA 0 = 3.25 CM
 DELTA* 0 = 1.26 CM
 THETA 0 = 0.083 CM
 TANAL INF = 21.6 N/M^2
 Q INF = 10400 W/M^2
 BE/M = 5.3EE+06
 CF INF = 0.99EE-03
 CH INF = 0.56EE-03

UPSTREAM BOUNDARY LAYER

Y (cm)	M	P / P INF	RHO / RHO INF	T / T INF	U / U INF	RMQU / RMQU INF	TT / TT INF
0.000	0.000	1.000	0.267	3.744	0.000	0.000	0.270
0.110	2.350	1.000	0.243	4.109	0.571	0.139	0.609
0.210	3.252	1.000	0.290	3.445	0.730	0.212	0.753
0.320	3.530	1.000	0.318	3.148	0.758	0.241	0.770
0.420	3.772	1.000	0.338	2.957	0.785	0.265	0.795
0.520	4.058	1.000	0.356	2.811	0.823	0.293	0.840
0.620	4.297	1.000	0.380	2.631	0.843	0.320	0.859
0.720	4.550	1.000	0.406	2.464	0.864	0.351	0.879
0.820	4.703	1.000	0.423	2.362	0.874	0.370	0.889
0.930	5.076	1.000	0.490	2.043	0.877	0.430	0.873
1.030	5.247	1.000	0.499	2.002	0.898	0.448	0.902
1.130	5.477	1.000	0.526	1.903	0.914	0.480	0.921
1.240	5.678	1.000	0.559	1.789	0.919	0.514	0.921
1.340	5.891	1.000	0.582	1.718	0.934	0.544	0.942
1.440	6.039	1.000	0.600	1.666	0.943	0.566	0.953
1.540	6.259	1.000	0.628	1.591	0.955	0.600	0.968
1.640	6.371	1.000	0.636	1.572	0.966	0.614	0.986
1.750	6.571	1.000	0.670	1.492	0.971	0.651	0.989
1.850	6.765	1.000	0.700	1.429	0.978	0.685	0.997
1.950	7.005	1.000	0.746	1.361	0.981	0.732	0.996
2.050	7.171	1.000	0.783	1.278	0.980	0.767	0.991
2.150	7.373	1.000	0.817	1.224	0.987	0.806	0.988
2.250	7.492	1.000	0.843	1.186	0.987	0.832	0.996
2.350	7.647	1.000	0.869	1.150	0.992	0.862	1.003
2.450	7.800	1.000	0.901	1.110	0.994	0.896	1.004
2.550	7.875	1.000	0.918	1.090	0.994	0.912	1.003
2.650	7.949	1.000	0.936	1.068	0.994	0.930	1.001
2.750	8.023	1.000	0.944	1.059	0.999	0.943	1.009
2.850	8.074	1.000	0.941	1.063	1.007	0.948	1.023
2.940	8.132	1.000	0.940	1.042	1.004	0.964	1.017
3.030	8.189	1.000	0.942	1.039	1.010	0.972	1.027
3.120	8.233	1.000	0.988	1.012	1.002	0.990	1.011
3.220	8.275	1.000	0.999	1.001	1.001	1.000	1.009
3.310	8.275	1.000	1.009	0.991	0.996	1.005	1.000

10 x 10 DEGREE SHOCK GENERATOR
 STREAMWISE CENTERLINE SURFACE PRESSURES ON FLAT PLATE

X (cm)	P/P INF
0.20	0.85
1.20	0.84
2.20	0.82
3.19	0.85
4.19	0.82
5.22	0.90
6.20	0.86
7.20	0.88
8.20	0.89
9.20	1.01
10.20	1.16
11.20	1.43
12.19	1.69
13.19	2.05
14.19	2.42
15.19	2.92
16.19	3.23
22.35	5.24
23.35	5.73
24.35	6.19
25.34	6.87
26.34	7.52
27.37	8.27
28.35	8.85
29.35	9.55
30.35	10.31
31.35	11.48
32.35	13.23
33.35	14.97
34.34	16.92
35.34	13.48
36.34	11.92
37.34	10.66
38.34	9.52

10 x 10 DEGREE SHOCK GENERATOR
 STREAMWISE CENTERLINE SURFACE HEAT TRANSFER ON FLAT PLATE
 FROM THERMOCOUPLES

X (cm)	Q/Q INF
4.03	1.02
5.01	1.00
6.02	1.02
7.00	0.99
7.96	1.01
8.95	1.01
9.95	0.98
10.96	0.91
12.98	1.09
13.98	1.39
14.98	1.73
16.98	2.34
17.99	2.37
18.99	2.61
20.03	2.74
22.50	3.21
23.48	3.48
24.49	3.79
25.47	4.04
26.43	4.47

27.42 4.94
28.42 5.21
29.43 5.59
31.45 6.38
32.45 7.21
33.45 8.02
35.71 7.33
36.46 6.48
37.46 5.84
38.50 5.28

15 x 15 DEGREE SHOCK GENERATOR
STREAMWISE CENTERLINE SURFACE PRESSURES ON FLAT PLATE

X (cm)	P/P INF
0.25	1.09
1.25	1.07
2.25	1.05
3.24	1.09
4.24	1.03
5.27	1.15
6.25	1.19
7.25	1.45
8.25	1.81
9.25	2.29
11.25	3.81
12.24	4.69
13.24	5.50
14.24	6.08
15.24	6.85
15.50	6.90
16.24	7.74
16.50	7.85
17.50	9.27
18.49	11.40
19.49	13.66
20.52	16.11
21.50	18.71
22.50	21.13
23.50	20.32
23.58	20.16
24.50	17.90
24.58	17.74
25.50	15.45
25.58	14.63
26.50	13.44
26.57	12.40
27.49	11.87
27.57	10.69
28.49	10.92
28.60	9.89
29.49	11.45
29.58	11.10
30.49	13.73
30.58	14.13
31.49	16.29
31.58	16.61
32.58	17.26
33.58	16.61
34.58	15.65
35.57	14.79
36.57	14.19
37.57	14.03
38.57	14.39
39.57	15.42

15 x 15 DEGREE SHOCK GENERATOR
STREAMWISE CENTERLINE SURFACE HEAT TRANSFER ON FLAT PLATE
FROM THERMOCOUPLES

X (cm)	Q/Q INF
2.00	0.74
2.98	0.87
3.99	0.78
4.97	0.75
5.93	0.78
6.92	0.74
7.92	0.58
8.93	0.59
10.95	1.76
11.95	2.60
12.95	3.42
14.00	4.02
14.98	4.38
15.21	4.60
15.96	4.68
15.99	4.72
16.96	5.20
16.97	5.16
17.93	6.22
18.00	6.36
18.92	7.63
19.92	8.90
20.93	10.28
22.95	12.09
23.95	11.69
24.95	10.28
27.21	7.14
27.96	6.31
28.96	6.14
30.00	6.88

TRANVERSE SURFACE PRESSURE, YAW ANGLE, AND HEAT TRANSFER DISTRIBUTION

10 x 10 DEGREE SHOCK GENERATOR
TRANVERSE SURFACE QUANTITIES, STATION 1

Pressures (X = 18.2 cm)

Z (cm)	P/P INF
0.32	3.56
0.41	3.76
0.62	3.40
0.71	3.68
1.30	3.05
1.41	3.31
1.60	2.87
1.71	3.18
2.30	2.65
2.40	2.74
2.60	2.85
2.70	2.69
3.30	3.58
3.40	3.37

EXCERPT ONLY
SEE DATA FILE
KUSOY4.DAT

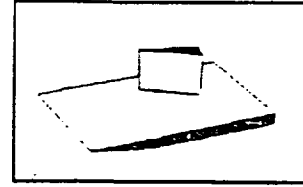
Ref.: 10-11

Author: Lee and Settles

Geometry: 3-D Fin

Mach number: 3, 4

Data: c_h



The Ph.D. research of Yeol Lee in the Penn State Gas Dynamics Laboratory involved measuring the heat transfer distributions beneath several sharp-fin interactions. The test conditions at Mach 3 and 4 and the fin angle-of-attack range were the same as those already documented here and in Ref. 1 for the Penn State series of sharp-fin experiments, which also includes surface pressures, surface flow directions, skin friction distributions, and flowfield density profiles. The present heat transfer data, listed in the file LEE.DAT, are thus supplementary data.

These heat transfer measurements required the development of a special heated-model technique for steady-state heat transfer in an otherwise near-adiabatic wind tunnel. The technique is discussed at length in Refs. 10 and 11. It was assumed, for simplicity of the instrumentation layout, that the quasiconical-flow approximation is valid for sharp-fin interactions (see, eg, Alvi and Settles, *AIAA Journal*, Vol. 30, Sept. 1992, pp. 2252-2258). 37 RTD heat transfer gages were thus arrayed in two adjacent circular arcs, centered at the fin leading edge location, with radii of 86.4 and 91.4 mm. The fin leading edge was 22.16 cm aft of the flat plate leading edge. The lateral spacing of the gages was such that their resolution was 2 degrees of the azimuthal interaction angle β over a β range of 6 to 78 degrees. the radii of the two gage rows was chosen to place them well outside the inception zones of all interaction cases studied.

A special feature of this dataset is the fact that adiabatic wall temperatures were actually measured rather than inferred, so the data are presented in terms of true heat transfer coefficients rather than dimensional heating rates. For more detail on this issue, please see Refs. 10 and 11.

```

*****
LEE-DAT
HEAT TRANSFER-
SHARP FIN SHOCK/BL INTERACTION DATA - PENN STATE GAS DYNAMICS LAB
*****
FREESTREAM CONDITIONS:
Mach number 3.03 (Mach 3)
stagnation pressure 120 psia (827 kPa)
stagnation temperature 295 K
Reynolds number 6.19E+07/in
Tt/Taw = 1.06 (roughly adiabatic)

Mach number 3.98 (Mach 4)
stagnation pressure 221 psia (1524 kPa)
stagnation temperature 293 K
Reynolds number 6.79E+07/in
Tt/Taw = 1.06 (roughly adiabatic)

INCOMING BOUNDARY LAYER PARAMETERS AT X=178 mm FROM FLAT-PLATE
LEADING EDGE

Mach 3 (actual freestream Mach number was 2.91 for this survey)
Delta = 3.02 mm
Delta* = 0.895 mm
Theta = 0.184 mm
Cf = 0.001520
Re(theta) = 9751

Mach 4 (actual freestream Mach number was 3.88 for this survey)
Delta = 2.87 mm
Delta* = 0.950 mm
Theta = 0.128 mm
Cf = 0.001325
Re(theta) = 9082

*****
REFERENCE NORMALIZERS
*****
MACH 3 CH INF = 0.86E-03
MACH 4 CH INF = 0.70E-03
*****
TABULATED DATA
NOTE: Beta values are measured from fin leading edge, not VCO
*****
MACH 3, 10DEG FIN INTERACTION
(max total uncertainty ± 10.9%)
BETA CH/ CH INF
( DEG) ( ) ( )
-----
57.8 1.000
55.9 0.930
53.7 0.974
51.8 0.983
47.7 0.990
41.4
39.5 0.922
33.2 0.940
31.2 0.970
29.0 0.998
27.0 1.172
24.9 1.361
22.9 1.508
20.7 1.727
18.7 1.911
16.6 2.045
12.5 1.963
*****

```

MACH 3, 160DEG FIN INTERACTION
(max total uncertainty ± 8.8%)

BETA (DEG)	CH/ ()	CH INF ()
57.8	1.000	
55.9	0.933	
53.7	0.978	
51.8	1.029	
47.7	1.024	
41.4	1.073	
39.5	1.052	
33.2	1.506	
31.2	1.764	
29.0	2.033	
27.0	2.456	
24.9	2.944	
22.9	3.237	
20.7	3.183	
18.7	2.864	

MACH 4, 20DEG FIN INTERACTION
(max total uncertainty ± 9.0%)

BETA (DEG)	CH/ ()	CH INF ()
57.8	1.000	
55.9	0.948	
53.7	0.976	
51.8	1.034	
47.7	1.177	
41.4	1.616	
39.5	1.455	
33.2	2.082	
31.2	2.492	
29.0	2.992	
27.0	4.227	
24.9	5.120	
22.9	4.452	

*****END OF FILE*****

MACH 3, 20DEG FIN INTERACTION
(max total uncertainty ± 12.8%)

BETA (DEG)	CH/ ()	CH INF ()
57.8	1.000	
55.9	0.940	
53.7	0.965	
51.8	1.032	
47.7	1.270	
41.4	1.686	
39.5	1.400	
33.2	2.438	
31.2	2.693	
29.0	3.939	
27.0	4.110	
24.9	3.455	
22.9	3.074	

MACH 4, 160DEG FIN INTERACTION
(max total uncertainty ± 9.5%)

BETA (DEG)	CH/ ()	CH INF ()
57.8	1.000	
55.9	0.921	
53.7	0.955	
51.8	0.976	
47.7	0.942	
41.4	1.081	
39.5	0.921	
33.2	1.325	
31.2	1.618	
29.0	1.795	
27.0	2.167	
24.9	2.643	
22.9	3.136	
20.7	4.081	
18.7	3.652	

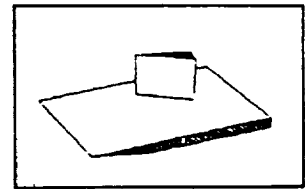
Ref.: 7-9

Author: Rodi and Dolling

Geometry: 3-D Fin

Mach number: 4.9

Data: p_{wall} , q_{wall} , surface-flow traces



In this experiment, a single sharp fin was used to generate an attached oblique shock wave which interacted with an incoming turbulent boundary layer formed along a flat plate test surface. The freestream Mach number was 4.90, the nominal total pressure was 315 psia (2.7 MPa) and the nominal total temperature was 760 degrees R (422 K). The freestream Reynolds number was $11.7 \times 10^6/\text{ft.}$ ($38.4 \times 10^6/\text{m}$). The model was at ambient temperature before each run. This combination of tunnel total temperature and model temperature resulted in a cool wall condition with a wall temperature/recovery temperature ratio of 0.8. Pitot pressure surveys were made of the incoming boundary layer just ahead of the fin and are included in this dataset.

Mean surface heat transfer (using Schmidt-Boelter gages) and surface pressure data (using pressure taps and a scani-valve) were measured on the test surface along spanwise rows for a range of fin angle of attack. The kerosene-lampblack surface tracer technique was used to visualize the distribution of the local shear stress direction. Data were collected at six different fin angles of attack (6-, 8-, 10-, 12-, 14- and 16-degrees). Additional heat transfer data were taken along conical rays from the virtual conical origin (VCO) for fin angles of attack of 8- and 15-degrees. The investigator reported an error band of $\pm 8\%$ for the heat transfer data and $\pm 3\%$ for the pressure data.

The X-dimension is the downstream direction measured from the fin leading edge. The Y-dimension is the cross-stream direction measured from the fin leading edge, in the spanwise direction towards the compression side of the fin. The Z-direction is the vertical direction above the flat plate test surface.

Three different rotatable instrumentation "plugs" were used to make the present measurements. The "plug" used for wall pressure measurements had four rows of taps which were initially oriented in the spanwise (Y) direction (see diagram 1). The spanwise pressure data for 6, 8, 10, 12, and 14 degree fin angles were taken on the two downstream tap rows (rows 3 and 4 in diagram 1). Then, by rotating the instrumentation plug clockwise, the pressure-tap rows were reoriented to better approximate conical cross-planes (see diagram 2). For 8 and 16-degree fin angles, the plug was rotated 15 and 30 degrees clockwise, respectively, from its initial position. For these two fin angles, three rows of pressure data (rows 2, 3, and 4) are tabulated for both spanwise and rotated orientations of the instrumentation plug.

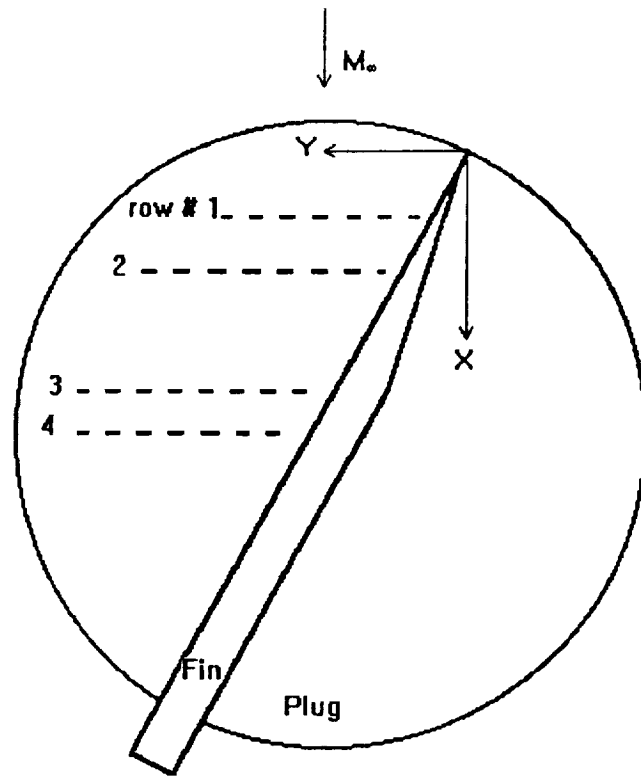


Diagram 1

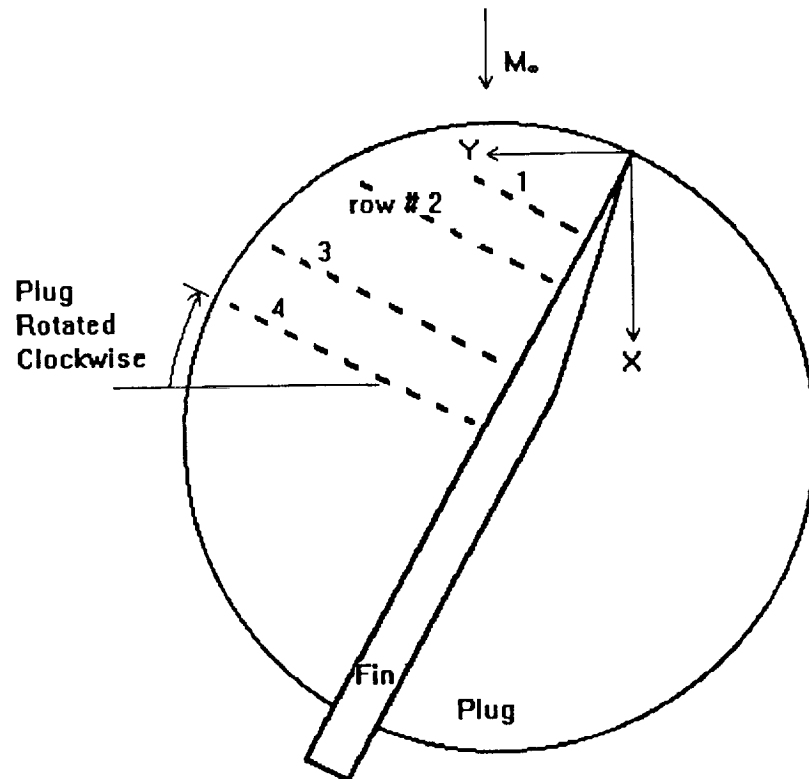


Diagram 2

 RODI.DAT

SHARP-FIN-GENERATED SHOCK WAVE/TURBULENT BOUNDARY
 LAYER INTERACTION DATA AT MACH 4.9, INCLUDING
 HEAT TRANSFER MEASUREMENTS

NOMENCLATURE

ALPHA = fin angle of attack, degrees
 BETA = conical angle in plane of test surface, degrees
 BETAO = angle of inviscid shock trace, degrees
 BETAU = upstream separation angle, degrees
 BETAST = primary separation angle, degrees
 BEVCO = beta angle from VCO, degrees
 BEFLE = beta angle from fin leading edge, degrees
 BLM = calculated local Mach number within boundary layer
 BLP = assumed static pressure within boundary layer, psia
 BLPO = measured total pressure within boundary layer, psia
 BLTO = assumed total temperature within boundary layer, F
 CN = Stanton number
 CHE = Stanton number at interaction edge
 DELTA = incoming boundary layer thickness, 0.25 in.
 DELT = local spanwise distance from fin to tap/transducer, inches
 LI = length of leading edge inception region, inches
 P = static wall pressure, psia
 PE = pressure at interaction edge, psia
 PO = tunnel total pressure, psia
 QDOT = wall heat transfer, BTU/ft²/sec
 R = radial distance from VCO, inches
 THETA = plug rotation angle, degree
 TO = tunnel total temperature, degrees F
 TU = wall temperature, degrees F
 VCO = virtual conical origin
 VCOB = VCO based upon the wall pressure distribution
 VCOB = VCO based upon the surface tracer patterns
 X = streamwise distance, inches
 XLE = streamwise distance from fin leading edge, inches
 Y = spanwise distance, inches
 YS = local spanwise distance from fin to inviscid shock, inches
 YS = spanwise distance from fin normalized by the VS
 YLE = spanwise distance from fin leading edge, inches
 Z = vertical distance above plate, inches

INCOMING BOUNDARY LAYER DOCUMENTATION

The incoming turbulent boundary layer Pitot profile was measured 19.3 in. (49.0 cm.) downstream of the flat plate leading edge and approximately 0.5 in. (1.27 cm.) upstream of the fin leading edge. Height above the wall (Z) and Pitot pressure (BLPO) were measured directly, the local static pressure distribution (BLP) was assumed to be linear as was the local total temperature distribution (BLTO). Local Mach number (BLM) was calculated.

INDEX	Z (in)	BLP (psia)	BLPO (psia)	BLM	BLTO (F)
1	0.57000E-01	0.752	7.300	2.68	260.15
2	0.59449E-01	0.751	7.340	2.69	260.51
3	0.63031E-01	0.750	7.816	2.78	261.05
4	0.66614E-01	0.748	8.140	2.84	261.77
5	0.69449E-01	0.748	8.507	2.91	262.31
6	0.72000E-01	0.747	8.689	2.94	262.67
7	0.75666E-01	0.745	8.959	2.99	263.39
8	0.82834E-01	0.743	9.082	3.02	264.65
9	0.86260E-01	0.742	9.452	3.08	265.37
10	0.89291E-01	0.741	9.873	3.16	265.91
11	0.96575E-01	0.739	10.556	3.27	267.17
12	0.10024E+00	0.738	10.920	3.33	267.71
13	0.10571E+00	0.736	10.990	3.35	268.79

14	0.10864E+00	0.735	11.413	3.42	269.33
15	0.11327E+00	0.733	11.853	3.49	270.05
16	0.11949E+00	0.731	12.788	3.63	271.13
17	0.12524E+00	0.729	13.017	3.67	272.21
18	0.13209E+00	0.727	13.538	3.75	273.47
19	0.13776E+00	0.725	13.819	3.80	274.55
20	0.14559E+00	0.723	15.084	3.98	275.81
21	0.15303E+00	0.721	15.722	4.07	277.25
22	0.15929E+00	0.718	15.911	4.10	278.33
23	0.16587E+00	0.716	16.796	4.22	279.41
24	0.17480E+00	0.713	17.622	4.34	281.03
25	0.18327E+00	0.711	18.086	4.40	282.65
26	0.19091E+00	0.708	18.507	4.46	283.91
27	0.19795E+00	0.706	18.986	4.53	285.17
28	0.20594E+00	0.703	19.479	4.60	286.61
29	0.21563E+00	0.700	19.537	4.61	288.41
30	0.22791E+00	0.696	20.320	4.72	290.57
31	0.23752E+00	0.693	20.407	4.74	292.19
32	0.24707E+00	0.690	20.581	4.77	294.17
33	0.25802E+00	0.686	20.610	4.79	296.15
34	0.26827E+00	0.683	20.668	4.81	297.77
35	0.28004E+00	0.679	20.712	4.83	299.93
36	0.30000E+00	0.673	20.726	4.85	303.35
37	0.31067E+00	0.670	20.755	4.87	305.33

SURFACE PRESSURE DATA

Below are the spanwise pressure measurements for fin angles of attack = 6-, 10-, 12- and 14-degrees. The 8- and 16-degree spanwise data are plotted in conical coordinates along with the conical cross-plane data later. PE is defined as the edge pressure value along each spanwise row.

ALPHA= 6.0					BETAO= 16.1				
X (in.)	Y (in.)	DELTA (in.)	YS (in.)	P (psia)	DELTA/YS	P (psia)	P/PE		
2.009	1.852	1.840	0.369	4.450	4.450	0.777	1.000		
2.009	1.712	1.500	0.369	4.070	3.690	0.770	0.991		
2.009	1.572	1.360	0.369	3.690	3.311	0.761	0.979		
2.009	1.432	1.220	0.369	3.311	2.931	0.766	0.986		
2.009	1.292	1.080	0.369	2.931	2.551	0.765	0.985		
2.009	1.152	0.940	0.369	2.551	2.171	0.822	1.058		
2.009	1.012	0.800	0.369	2.171	1.792	0.907	1.167		
2.009	0.872	0.660	0.369	1.792	1.412	1.001	1.288		
2.009	0.732	0.520	0.369	1.412	1.032	1.032	1.328		
2.009	0.592	0.380	0.369	1.032	0.652	1.066	1.372		
2.009	0.452	0.240	0.369	0.652	0.273	1.142	1.470		
2.009	0.312	0.100	0.369	0.273	0.273	1.244	1.601		
2.378	1.852	1.602	0.436	3.670	3.350	0.755	1.000		
2.378	1.712	1.462	0.436	3.350	3.029	0.759	1.005		
2.378	1.572	1.322	0.436	3.029	2.708	0.761	1.008		
2.378	1.432	1.182	0.436	2.708	2.387	0.768	1.017		
2.378	1.292	1.042	0.436	2.387	2.066	0.827	1.095		
2.378	1.152	0.902	0.436	2.066	1.745	0.911	1.207		
2.378	1.012	0.762	0.436	1.745	1.425	0.976	1.293		
2.378	0.872	0.622	0.436	1.425	1.104	1.051	1.392		
2.378	0.732	0.482	0.436	1.104	0.783	1.070	1.417		
2.378	0.592	0.342	0.436	0.783	0.462	1.178	1.560		
2.378	0.452	0.202	0.436	0.462	0.369	1.244	1.601		
2.009	1.642	1.330	0.369	3.880	3.501	0.764	0.992		
2.009	1.502	1.290	0.369	3.501	3.121	0.792	1.029		
2.009	1.362	1.150	0.369	3.121	2.741	0.792	1.029		
2.009	1.222	1.010	0.369	2.741	2.361	0.867	1.126		
2.009	1.082	0.870	0.369	2.361	1.862	1.034	1.363		
2.009	0.802	0.590	0.369	1.862	1.222	1.057	1.373		
2.009	0.662	0.450	0.369	1.222	0.842	1.110	1.442		
2.009	0.522	0.310	0.369	0.842					

ALPHA= 10.0									
X	Y	DEL	YS	DEL/YS	P	P/PE	DEL/YS	YS	P
(in.)	(in.)	(in.)	(in.)	(in.)	(psia)			(in.)	(psia)
2.009	0.382	0.170	0.369	0.462	1.224	1.590		0.362	0.983
2.009	0.242	0.030	0.369	0.083	1.298	1.686	3.751	0.362	0.745
2.378	1.842	1.392	0.436	3.189	0.770	1.000	3.364	0.362	0.768
2.378	1.502	1.392	0.436	2.868	0.764	0.992	2.977	0.362	1.013
2.378	1.362	1.112	0.436	2.548	0.809	1.051	2.590	0.362	0.921
2.378	1.222	0.972	0.436	2.227	0.881	1.144	2.203	0.362	1.445
2.378	1.082	0.832	0.436	1.906	0.961	1.248	1.817	0.362	1.587
2.378	0.942	0.692	0.436	1.585	1.011	1.313	1.430	0.362	1.609
2.378	0.802	0.552	0.436	1.264	1.048	1.361	1.043	0.362	1.217
2.378	0.662	0.412	0.436	0.943	1.056	1.371	0.656	0.362	1.606
2.378	0.522	0.272	0.436	0.623	1.140	1.481	0.289	0.362	1.243
2.378	0.382	0.132	0.436	0.302	1.274	1.655	2.114	0.362	1.559
							2.789	0.362	2.057
							3.309	0.362	1.000
							0.755	0.362	1.026
							0.863	0.362	1.173
							1.040	0.362	1.413
							2.656	0.362	1.541
							2.330	0.362	1.134
							2.003	0.362	1.211
							1.677	0.362	1.645
							1.350	0.362	1.268
							1.024	0.362	1.896
							0.697	0.362	1.633
							0.371	0.362	1.220
							3.944	0.362	1.658
							0.773	0.362	2.042
							0.842	0.362	2.779
							1.012	0.362	1.503
							1.042	0.362	2.945
							1.172	0.362	1.000
							1.265	0.362	0.764
							1.354	0.362	3.558
							1.447	0.362	0.773
							1.534	0.362	0.842
							1.623	0.362	1.012
							1.718	0.362	1.171
							1.809	0.362	1.344
							1.899	0.362	1.526
							1.995	0.362	1.712
							2.089	0.362	1.904
							2.187	0.362	2.099
							2.287	0.362	2.299
							2.384	0.362	2.499
							2.482	0.362	2.699
							2.581	0.362	2.899
							2.679	0.362	3.099
							2.779	0.362	3.299
							2.877	0.362	3.499
							2.977	0.362	3.699
							3.077	0.362	3.899
							3.177	0.362	4.099
							3.277	0.362	4.299
							3.377	0.362	4.499
							3.477	0.362	4.699
							3.577	0.362	4.899
							3.677	0.362	5.099
							3.777	0.362	5.299
							3.877	0.362	5.499
							3.977	0.362	5.699
							4.077	0.362	5.899
							4.177	0.362	6.099
							4.277	0.362	6.299
							4.377	0.362	6.499
							4.477	0.362	6.699
							4.577	0.362	6.899
							4.677	0.362	7.099
							4.777	0.362	7.299
							4.877	0.362	7.499
							4.977	0.362	7.699
							5.077	0.362	7.899
							5.177	0.362	8.099
							5.277	0.362	8.299
							5.377	0.362	8.499
							5.477	0.362	8.699
							5.577	0.362	8.899
							5.677	0.362	9.099
							5.777	0.362	9.299
							5.877	0.362	9.499
							5.977	0.362	9.699
							6.077	0.362	9.899
							6.177	0.362	10.099
							6.277	0.362	10.299
							6.377	0.362	10.499
							6.477	0.362	10.699
							6.577	0.362	10.899
							6.677	0.362	11.099
							6.777	0.362	11.299
							6.877	0.362	11.499
							6.977	0.362	11.699
							7.077	0.362	11.899
							7.177	0.362	12.099
							7.277	0.362	12.299
							7.377	0.362	12.499
							7.477	0.362	12.699
							7.577	0.362	12.899
							7.677	0.362	13.099
							7.777	0.362	13.299
							7.877	0.362	13.499
							7.977	0.362	13.699
							8.077	0.362	13.899
							8.177	0.362	14.099
							8.277	0.362	14.299
							8.377	0.362	14.499
							8.477	0.362	14.699
							8.577	0.362	14.899
							8.677	0.362	15.099
							8.777	0.362	15.299
							8.877	0.362	15.499
							8.977	0.362	15.699
							9.077	0.362	15.899
							9.177	0.362	16.099
							9.277	0.362	16.299
							9.377	0.362	16.499
							9.477	0.362	16.699
							9.577	0.362	16.899
							9.677	0.362	17.099
							9.777	0.362	17.299
							9.877	0.362	17.499
							9.977	0.362	17.699
							10.077	0.362	17.899
							10.177	0.362	18.099
							10.277	0.362	18.299
							10.377	0.362	18.499
							10.477	0.362	18.699
							10.577	0.362	18.899
							10.677	0.362	19.099
							10.777	0.362	19.299
							10.877	0.362	19.499
							10.977	0.362	19.699
							11.077	0.362	19.899
							11.177	0.362	20.099
							11.277	0.362	20.299
							11.377	0.362	20.499
							11.477	0.362	20.699
							11.577	0.362	20.899
							11.677	0.362	21.099
							11.777	0.362	21.299
							11.877	0.362	21.499
							11.977	0.362	21.699
							12.077	0.362	21.899
							12.177	0.362	22.099
							12.277	0.362	22.299
							12.377	0.362	22.499
							12.477	0.362	22.699
							12.577	0.362	22.899
							12.677	0.362	23.099
							12.777	0.362	23.299
							12.877	0.362	23.499
							12.977	0.362	23.699
							13.077	0.362	23.899
							13.177	0.362	24.099
							13.277	0.362	24.299
							13.377	0.362	24.499
							13.477	0.362	24.699
							13.577	0.362	24.899
							13.677	0.362	25.099
							13.777	0.362	25.299
							13.877	0.362	25.499
							13.977	0.362	25.699
							14.077	0.362	25.899
							14.177	0.362	26.099
							14.277	0.362	26.299
							14.377	0.362	26.499
							14.477	0.362	26.699
							14.577	0.362	26.899
							14.677	0.362	27.099
							14.777	0.362	27.299
							14.877	0.362	27.499
							14.977	0.362	27.699
							15.077	0.362	27.899
							15.177	0.362	28.099
							15.277	0.362	28.299
							15.377	0.362	28.499
							15.477	0.362	28.699
							15.577	0.362	28.899
							15.677	0.362	29.099
							15.777	0.362	29.299
							15.877	0.362	29.499
							15.977	0.362	29.699
							16.077	0.362	29.899
							16.177	0.362	

***** Surface flow Visualization Results *****

The following table lists the results from the SFVP study. The inception length (LI), and angles of upstream influence (BETAU) and primary separation line (BETAS) are presented below (DELTA=0.25 in.).

ALPHA (deg)	LI/DELTA	BETAU (deg)	BETAS (deg)
6.0	4.44	22.	20.
8.0	4.28	26.	23.
10.0	3.76	29.	27.
12.0	3.16	34.	31.
14.0	2.72	36.	34.
15.0 (Gibson)	N/A	36.	34.

Using the primary separation line (clearly seen on the SFVP) and the inviscid shock trace (calculated analytically) a VCD can be determined from the intersection of these two features.

Location of VCDv from Primary Separation Line and Inviscid Shock Wave

ALPHA (deg)	X/DELTA	Y/DELTA
6.0	-6.64	-2.04
8.0	-5.48	-1.76
10.0	-4.52	-1.60
12.0	-3.48	-1.36
14.0	-3.24	-1.40

The locations of the virtual conical origin of the pressure fields, obtained by matching features of the various cross plane distributions and extrapolating back to the VCDp, are list below.

Location of VCDp from Pressure Distributions

ALPHA (deg)	X/DELTA	Y/DELTA
8.0	-3.55	-1.44
16.0	-1.90	-0.91

***** Spanwise Heat Transfer Data *****

Next are given the spanwise heat transfer measurements for fin angles of attack of 6-, 8-, 10-, 12-, 14- and 16-degrees. Data from four runs were combined to produce a distribution with increased spatial resolution.

ALPHA= 6.000 BETA0= 16.100

TO= 299.04(F)	CHE= 0.00050	X	Y	DELTA	YS	DELY/YS	TU	QDOT	CH	CH/CHE
(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(F)	(BTU/FT ² -2sec)		
2.009	1.886	1.675	0.369	4.54	78.53	0.696	0.00050	1.000		
2.009	1.324	1.113	0.369	3.02	82.59	0.705	0.00052	1.040		
2.009	1.043	0.832	0.369	2.26	81.23	0.790	0.00058	1.160		
2.009	0.762	0.551	0.369	1.49	92.07	0.841	0.00066	1.320		
2.009	0.481	0.270	0.369	0.73	87.39	1.346	0.00103	2.060		

TO= 301.52(F)	CHE= 0.00050	X	Y	DELTA	YS	DELY/YS	TU	QDOT	CH	CH/CHE
(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(F)	(BTU/FT ² -2sec)		

TO= 303.45(F)	CHE= 0.00049	X	Y	DELTA	YS	DELY/YS	TU	QDOT	CH	CH/CHE
(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(F)	(BTU/FT ² -2sec)		
2.009	1.746	1.535	0.369	4.16	103.76	0.577	0.00048	0.980		
2.009	1.465	1.254	0.369	3.40	103.72	0.599	0.00050	1.020		
2.009	1.184	0.973	0.369	2.64	105.50	0.685	0.00058	1.184		
2.009	0.903	0.692	0.369	1.88	105.37	0.756	0.00064	1.306		
2.009	0.622	0.411	0.369	1.11	114.88	0.843	0.00077	1.571		
2.009	0.341	0.130	0.369	0.35	98.14	1.281	0.00103	2.102		

TO= 301.94(F)	CHE= 0.00050	X	Y	DELTA	YS	DELY/YS	TU	QDOT	CH	CH/CHE
(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(F)	(BTU/FT ² -2sec)		
2.009	1.816	1.605	0.369	4.35	86.19	0.668	0.00050	1.000		
2.009	1.535	1.324	0.369	3.59	86.80	0.679	0.00051	1.020		
2.009	1.254	1.043	0.369	2.83	89.36	0.724	0.00055	1.100		
2.009	0.973	0.762	0.369	2.07	88.76	0.767	0.00058	1.160		
2.009	0.692	0.481	0.369	1.30	97.39	0.828	0.00067	1.340		
2.009	0.411	0.200	0.369	0.54	86.88	1.411	0.00106	2.120		

ALPHA= 8.000 BETA0= 17.600

TO= 301.60(F)	CHE= 0.00050	X	Y	DELTA	YS	DELY/YS	TU	QDOT	CH	CH/CHE
(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(F)	(BTU/FT ² -2sec)		
2.007	1.967	1.685	0.362	4.65	95.60	0.624	0.00050	1.000		
2.007	1.405	1.123	0.362	3.10	97.08	0.633	0.00051	1.020		
2.007	1.124	0.842	0.362	2.32	95.74	0.766	0.00061	1.220		
2.007	0.843	0.561	0.362	1.55	108.30	0.804	0.00071	1.620		
2.007	0.562	0.280	0.362	0.77	101.39	1.413	0.00118	2.360		

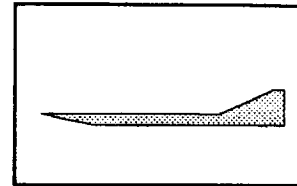
TO= 300.61(F)	CHE= 0.00049	X	Y	DELTA	YS	DELY/YS	TU	QDOT	CH	CH/CHE
(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(F)	(BTU/FT ² -2sec)		
2.007	1.756	1.474	0.362	4.07	97.62	0.672	0.00055	1.122		
2.007	1.475	1.193	0.362	3.29	96.08	0.613	0.00049	1.000		
2.007	1.194	0.912	0.362	2.52	108.52	0.802	0.00071	1.449		
2.007	0.913	0.631	0.362	1.76	101.06	0.765	0.00064	1.306		
2.007	0.632	0.350	0.362	0.97	104.13	1.179	0.00101	2.061		

TO= 303.74(F)	CHE= 0.00048	X	Y	DELTA	YS	DELY/YS	TU	QDOT	CH	CH/CHE
(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(F)	(BTU/FT ² -2sec)		
2.007	1.817	1.535	0.362	4.24	92.20	0.629	0.00048	1.000		
2.007	1.536	1.254	0.362	3.46	94.04	0.654	0.00051	1.042		
2.007	1.255	0.973	0.362	2.69	95.94	0.788	0.00062	1.292		
2.007	0.974	0.692	0.362	1.91	95.73	0.826	0.00065	1.354		
2.007	0.693	0.411	0.362	1.13	106.70	1.096	0.00094	1.958		
2.007	0.412	0.130	0.362	0.36	89.25	1.699	0.00128	2.667		

TO= 303.88(F)	CHE= 0.00048	X	Y	DELTA	YS	DELY/YS	TU	QDOT	CH	CH/CHE
(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(F)	(BTU/FT ² -2sec)		
2.007	1.897	1.615	0.362	4.44	98.44	0.603	0.00048	1.000		
2.007	1.616	1.334	0.362	3.68	97.71	0.613	0.00049	1.021		
2.007	1.335	1.053	0.362	2.91	100.58	0.676	0.00055	1.146		
2.007	1.054	0.772	0.362	2.13	100.04	0.727	0.00059	1.229		
2.007	0.773	0.491	0.362	1.36	108.73	0.934	0.00081	1.688		
2.007	0.492	0.210	0.362	0.58	99.36	1.565	0.00126	2.625		

EXCERPT-
SEE DATA
FILE:
R001.DAT

Author: Settles, G. S., *et al*
Geometry: 2-D Compression Corner
Mach number: 3
Data: p_{wall} , c_f , mean flowfield pitot surveys



Settles, G. S., Fitzpatrick, T. J. and Bogdonoff, S. M., "Detailed Study of Attached and Separated Compression Corner Flowfields in High Reynolds Number Supersonic Flow," *AIAA Journal*, Vol. 17, No. 6, June 1979, pp. 579.

Settles, G.S., Gilbert, R.B. and Bogdonoff, S.M., "Data Compilation For Shock Wave/Turbulent Boundary Layer Interaction Experiments On Two-Dimensional Compression Corners," *Princeton University Report 1489-MAE*, Princeton Univ. 1980.

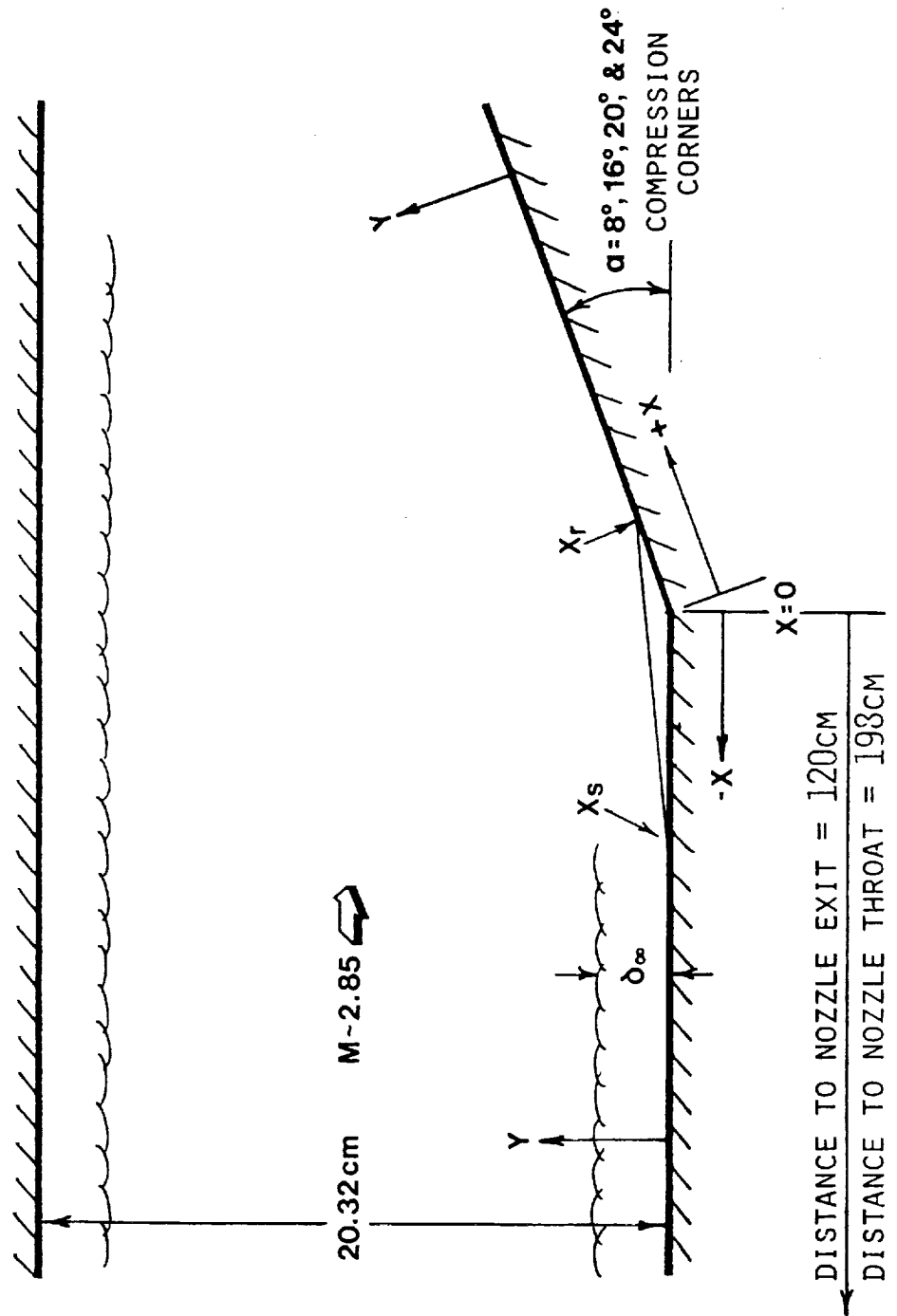
The data consist of both mean surveys of flow properties before and after two-dimensional compression corners at Mach numbers in the vicinity of 2.9. Compression corner angles of 8, 16, 20, and 24 degrees span the range from attached flow to large boundary-layer separation. Two-dimensionality of the experiments was demonstrated by studies of spanwise oil-flow patterns for all but the largest compression corner angle, where significant 3-D perturbations were observed.

The mean data include surface pressure and skin friction distributions, as well as pitot and static pressure distributions from which velocity and Mach number were deduced (the total temperature distribution through these interactions was nearly constant). *Note: the fluctuating data corresponding to this mean dataset were presented in the file SMITS.DAT in Ref. 1 and remain unchanged. The present correction and re-tabulation covers only the mean-flow data. It is given as file SETTLES2.DAT on the accompanying diskette.*

All units in the tables are SI. The x-coordinate is defined in the streamwise direction along the wind tunnel floor and compression corner surfaces. Thus locations upstream of the corner have negative x-values and those downstream have positive values. The compression corners were all located at 1.205 m downstream of the wind tunnel nozzle exit with the exception of the 24 degree corner for hot-wire measurements only, which was located 1.17 m downstream of the nozzle exit. The y-coordinate is measured upward from the test surface with its origin at that surface. The origin of the z-coordinate is on the wind tunnel centerline. It is taken positive to the left when looking downstream. See the diagram of the flow configuration reproduced below.

Users are encouraged to consult the cited references for detailed discussions of the data and their significance, which are beyond the present scope, as well as estimates of the various errors and discrepancies which serve to set confidence limits upon the data.

SCHEMATIC DIAGRAM OF FLOW CONFIGURATION
 COMPRESSION CORNERS
 PRINCETON HIGH REYNOLDS NUMBER SUPERSONIC WIND TUNNEL
 TUNNEL WIDTH= 20.32 CM



*****MEAN-FLOW DATA*****
 SETTLES at al 2-D COMPRESSION CORNER INTERACTIONS

DEFINITIONS

alpha = compression corner angle, degrees
 CF = TAUWAL/RHOINF*UREF**2
 DELTA = boundary layer thickness, cm
 DELTA* = boundary layer displacement thickness, cm
 Href = Nominal incoming freestream Mach number
 PINF = Freestream static pressure, N/m**2
 P11 = Freestream total pressure, N/m**2
 Pw, Pwall = Wall pressure, N/m**2
 RHOINF = Freestream density, kg/m**2
 SECTION = Wind tunnel test section (1, 2, or 3; see source document)
 TAUWAL = Wall shear stress, N/m**2
 THETA = boundary layer momentum thickness, cm
 TT1 = Freestream total temperature, K
 TWINF = Flat-plate wall temperature ahead of interaction
 UREF = Freestream velocity, m/s
 X = Streamwise coordinate, zero at corner, measured along flat plate or ramp, m
 Y = Vertical coordinate, zero at model, measured along direction of survey line, m
 Z = Spanwise distance from model centerline, m

NOMINAL FREESTREAM AND INCOMING CONDITIONS

 All cases: freestream Re/m = 6.3E+07, Distance from nozzle to compression corner = 198 cm.

alpha	8	16	20	24
Href	2.87	2.85	2.85	2.84
PINF	2.30E+04	2.29E+04	2.32E+04	2.36E+04 N/m**2
TT1	280	268	258	262 K
TWINF	291	282	274	276 K
UREF	592	576	562	569 m/s
DELTA	2.6	2.6	2.5	2.3 cm
DELTA*	0.67	0.63	0.66	0.61 cm
THETA	0.13	0.13	0.13	0.12 cm

*****UNCERTAINTIES*****

PWALL ± 2%
 CFINF ± 15%
 (PROFILE DATA)
 U ± 5%
 P ± 4%
 M ± 3%

*****SURFACE-PRESSURE DISTRIBUTIONS*****

8 DEGREE RAMP			16 DEGREE RAMP		
X, cm	Pwall/PINF	X, cm	Pwall/PINF		
-2.54	1.00	-5.08	1.01		
-1.02	1.00	-3.81	1.00		
-0.51	1.00	-2.22	1.00		
-0.25	1.02	-1.91	1.00		
-0.24	1.03	-1.58	1.00		
-0.18	1.06	-1.27	1.00		
-0.12	1.13	-0.95	1.00		
-0.06	1.22	-0.87	1.03		
0.0	1.35	-0.79	1.05		
0.06	1.42	-0.71	1.08		
0.12	1.45	-0.64	1.18		
0.18	1.46	-0.56	1.30		
0.24	1.48	-0.48	1.40		
0.25	1.47	-0.40	1.50		

20 DEGREE RAMP			24 DEGREE RAMP		
X, cm	Pwall/PINF	X, cm	Pwall/PINF		
0.38	1.47	-0.32	1.57		
0.51	1.47	-0.24	1.63		
0.76	1.48	-0.16	1.68		
1.02	1.49	-0.08	1.72		
1.52	1.52	0.00	1.77		
2.03	1.55	0.08	1.85		
2.54	1.57	0.16	1.90		
3.56	1.60	0.24	1.94		
4.57	1.65	0.32	2.02		
5.08	1.67	0.40	2.04		
5.59	1.68	0.48	2.08		
6.60	1.70	0.56	2.12		
7.62	1.70	0.64	2.16		
8.64	1.71	0.95	2.23		
9.65	1.72	1.27	2.30		
10.16	1.73	2.22	2.45		
10.67	1.73	3.81	2.65		
12.70	1.74	5.72	2.78		
12.70	1.74	7.62	2.82		
15.24	1.76	11.43	2.85		

20 DEGREE RAMP			24 DEGREE RAMP		
X, cm	Pwall/PINF	X, cm	Pwall/PINF		
-5.08	0.99	-7.62	1.02		
-3.81	1.00	-6.35	1.01		
-2.70	1.00	-5.08	1.02		
-2.38	1.03	-4.57	1.04		
-2.22	1.07	-4.06	1.26		
-2.06	1.16	-3.56	1.64		
-1.91	1.32	-3.30	1.69		
-1.75	1.44	-2.79	1.80		
-1.59	1.56	-2.29	1.90		
-1.43	1.60	-1.78	1.99		
-1.27	1.67	-1.27	2.07		
-1.11	1.71	-0.76	2.15		
-0.95	1.76	-0.25	2.23		
-0.79	1.78	0.0	2.28		
-0.64	1.82	0.51	2.44		
-0.56	1.85	1.02	2.59		
-0.48	1.87	2.54	3.09		
-0.40	1.89	3.05	3.29		
-0.32	1.90	4.57	3.68		
-0.24	1.91	5.08	3.73		
-0.16	1.94	6.10	3.80		
-0.08	1.97	7.11	3.89		
0.00	2.00	8.64	3.97		
0.08	2.02	9.65	4.07		
0.16	2.06	10.67	4.17		
0.24	2.12	11.68	4.26		
0.48	2.20	12.70	4.25		
0.64	2.30				
0.95	2.42				
1.27	2.53				
2.22	2.83				
3.81	3.15				
5.72	3.32				
7.62	3.43				
11.43	3.56				

*****SKIN FRICTION DATA*****

ALL UNITS ARE SYSTE INTERNATIONAL
 FREESTREAM VALUES MAY NOT EXACTLY MATCH NOMINAL VALUES CITED

ALPHA=8 DEG RAMP, Z = -0.0127, PRESTON TUBE DIA. = 8.130E-04 m

X	PT1	TT1	TAUWAL	CF
-2.540E-02	6.894E+05	2.780E+02	1.335E+02	1.003E-03

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7. PERFORMING ORGANIZATION NAME(S) AND ADDRESS(ES) Department of Mechanical Engineering Penn State University University Park, PA 16802		8. PERFORMING ORGANIZATION REPORT NUMBER A-94078		
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13. ABSTRACT (Maximum 200 words) An effort was begun in 1989 at the Penn State University Gas Dynamics Laboratory to perform a critical review of the available hypersonic data and to assemble a selected database for purposes of computational fluid dynamics (CFD) code validation and turbulence modeling. The effort was sponsored by the National Aero-Space Plane (NASP) Program through NASA Ames Research Center, and was a part of an overall task to develop compressible turbulence models. NASA CR-177577, a database report on hypersonic shock wave/turbulent boundary-layer interactions, was the product of phase 1 of that effort. Phase 2 produced a similar database, NASA CR-177610, covering the topics of attached hypersonic boundary layers in pressure gradients and compressible turbulent mixing layers. The present report represents the result of the third and final phase: namely, recent additions and corrections to the hypersonic shock wave/turbulent boundary-layer interaction database originally given in NASA CR-177577. Seven new datasets are presented, consisting of supersonic and hypersonic experiments using single- and double-fin shock wave generators and a compression corner shock wave generator. Finally, typographical and other errors occurring in NASA CR-177577 have been corrected in a section of the present report, including clarification of some issues which were raised with regard to the original database by users in the interim.				
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